DEPARTMENT OF TRANSPORTATION
FEDERAL AVIATION ADMINISTRATION

A3EU
Revision 28
Raytheon

DH.125 Series 1A	HS.125 Series 401B
HS.125 Series 1B	HS.125 Series 403A(C)
DH.125 Series 1A-522	HS.125 Series 403B
HS.125 Series 1B-522	HS.125 Series F400B
DH.125 Series 1A/R-522	HS.125 Series F403B
HS.125 Series 1B/R-522	BH.125 Series 600A
DH.125 Series 1A/S-522	HS.125 Series 600A
HS.125 Series 1B/S-522	HS.125 Series 600B
DH.125 Series 3A	HS.125 Series 600B/1
HS.125 Series 3B	HS.125 Series 600B/2
DH.125 Series 3A/R	HS.125 Series 600B/3
HS.125 Series 3B/R	HS.125 Series F600B
DH.125 Series 3A/RA	HS.125 Series 700A
HS.125 Series 3B/RB	HS.125 Series 700B
HS.125 Series 3B/RC	BAe.125 Series 800A
HS.125 Series F3B	BAe.125 Series 800A (C-29A)
HS.125 Series F3B/RA	BAe.125 Series 800A (U-125)
BH.125 Series 400A	BAe.125 Series 800B
DH.125 Series 400A	BAe.125 Series 1000A
HS.125 Series 400B	BAe.125 Series 1000B
HS.125 Series 400B/1	Hawker 800
	Hawker 800 (U-125A)
	Hawker 1000
	Hawker 800XP

April 29, 2002

TYPE CERTIFICATE DATA SHEET NO. A3EU

This Data Sheet, which is part of Type Certificate No. A3EU prescribes conditions and limitations under which the product for which the Type Certificate was issued meets the airworthiness requirements of the Federal Aviation Regulations.

Type Certificate Holder: Raytheon Aircraft Company
9709 East Central
Wichita, Kansas 67206

Type Certificate A3EU was transferred from Raytheon Corporate Jets Inc., 3 Bishop Square, St. Albans Road West, Hatfield, Hertfordshire AL 10 9NE, United Kingdom, to Raytheon Aircraft Company (RAC) on August 1, 1995. Coincident with this transfer, the Federal Aviation Administration (FAA) has accepted the status of State of Design and State of Manufacture as defined by Annex 8 to the Convention on International Civil Aviation. Prior to August 1, 1995, products identified under Type Certificate A3EU were approved by the FAA in accordance with the Federal Aviation Regulation appropriate to Imported Products (FAR 21.29). Effective August 1, 1995 and after, these products are to be considered domestic products for the purpose of certification, and Federal Aviation Regulation 21.21 becomes appropriate.

Effective May 28, 1999, certain models identified as “B” series that had been previously certified by the UK Civil Aviation Authority were added to Type Certificate A3EU and this Data Sheet. The process for type certification of these aircraft is considered analogous to issuance of export airworthiness approvals, with exceptions, as allowed under 14 CFR 21.325(c).

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Under that section the requirements that are not met and the differences in configuration, if any, between the product to be exported and the related type certificated product, are listed on the export airworthiness approval as exceptions. The UK certificated "B" series aircraft can be considered to be the US approved type certificated "A" series with exceptions. The UK approved "B" series are eligible to receive FAA airworthiness certificates and registration for operation in the United States as a US approved "B" series when modified to comply with US standards (i.e. the modification eliminates the exception; see NOTE 50) and when all Airworthiness Directives applicable to the equivalent "A" series have been incorporated.

The box in the upper right corner of page 1 identifies the FAA Approved Series and Models. The FAA has accepted the responsibility for the promulgation to International Civil Aviation Organization (ICAO) Contracting States of airworthiness information for all such products in accordance with Annex 8. The Type Certificate Holder designated in this data sheet holds Type Design authority for the production of data associated with all such products.

I. Hawker Siddeley Model DH.125 Series 1A (Transport Aircraft), Approved September 25, 1964 (See NOTE 14).
Hawker Siddeley Model HS.125 Series 1B (Transport Aircraft), Approved May 28, 1999 (See NOTES 14 and 52).

Engines	2 Bristol Siddeley Viper 521 turbine engines.	
Fuel	Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22 Type 2, ASTM D.1655 Jet B. (See NOTE 4).	
Engine Limits	Take-off static thrust, standard day, sea level conditions (unrestricted) lbs.	3,120
	Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,120
	Maximum permissible engine rotor operating speed	100% (13,760 r.p.m.)
	Maximum permissible turbine outlet gas temperature:	
	Take-off (unrestricted)	695°C
	Maximum continuous	695°C
	Maximum for acceleration	695°C
	Starting maximum gas temperature	800°C
	Maximum permissible oil inlet temperature:	
	Continuous operation (See NOTE 13)	125°C
	Maximum permissible air bleed extraction of primary engine airflow	7.5%
Airspeed Limits (IAS)	V_{MO} (Maximum operating) from sea level to 26,800 feet	290 knots
	M_{MO} (Maximum operating) 26,800 ft. and above	0.735 M
	V_A (Maneuvering)	
	Sea level	181 knots
	10,000 ft.	182 knots
	20,000 ft.	183 knots
	30,000 ft.	195 knots
	40,000 ft.	212 knots
	Straight line variation between points shown.	
	V_{FE} (Flap speeds)	
	Deflection	
	15°	210 knots
	25°	160 knots
	50° or 45° (See NOTE 23)	145 knots

Airspeed Limits (IAS)(Cont.)

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	93 knots
V_{MCG} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	94 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage lower skin immediately aft of the equipment bay access hatch.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps retracted)

	Fwd of Datum				Aft of Datum			
	In-Flight		T.O. & Land		Autopilot disengaged		Autopilot engaged	
Wt. Lbs.	% SMC	In.	% SMC	In.	% SMC	In.	% SMC	In.
21,200	22.40	0.55	23.60	(0.54)*	33.80	9.80	32.60	8.70
20,550	-	-	-	-	34.30	10.20	33.00	9.00
19,000	-	-	-	-	34.10	9.95	32.80	8.82
17,800	18.80	3.78	20.20	2.52	-	-	-	-
16,800	-	-	-	-	33.10	9.10	-	-
16,600	-	-	-	-	-	-	31.60	7.80
13,000	18.00	4.51	20.00	2.71	-	-	-	-
12,350	-	-	-	-	37.50	13.10	35.50	11.30
12,100	18.00	4.51	20.00	2.71		-	-	-
10,800	-	-	26.00	(2.71)*	37.50	13.10	35.50	11.30

*(Aft of Datum)

Straight line variation between weights

Item (Extending)	Moment Change In Lbs.
Wing Flaps 15^o	+ 538
25^o	+ 879
50^o or 45^o (See NOTE 23)	+1,593
Main Landing Gear	- 1,800
Nose Landing Gear	+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum Ramp Weight	21,200 lbs. (See NOTE 12)
Maximum Brake release weight	21,200 lbs.
Maximum Landing Weight	19,550 lbs.
Maximum Zero Fuel Weight	13,000 lbs. (See NOTE 12)

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Total	1,231.5	8210	5.79

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible

25013, 25014, 25016 through 25023, 25025 through 25039, 25042, 25043, 25046, 25047, 25051, 25052, 25053, 25057, 25058, 25060, 25063 through 25068, 25070, 25073 through 25075, 25078 through 25080, 25082 through 25110

II. Hawker Siddeley Model DH.125 Series 1A-522 (Transport Aircraft), Approved February 3, 1966. (See NOTE 14)
Hawker Siddeley Model HS.125 Series 1B-522 (Transport Aircraft), Approved May 28, 1999. (See NOTES 14 and 52).

(The DH.125 Series 1A-522 and HS.125 Series 1B-522 aircraft differs from the DH.125 Series 1A and the HS.125 Series 1B aircraft, respectfully, in the following major features: (i) Introduction of Bristol Siddeley Viper 522 engines (ii) values of M_{MO} increased and V_{MO} decreased.)

Engines

2 Bristol Siddeley Viper 522 turbine engines.

Fuel

Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22 Type 2, ASTM D.1655 Jet B. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
Maximum permissible engine rotor Operating speed (5 minutes maximum)	100% (13,760 r.p.m.)

Engine Limits (cont.)

Maximum permissible turbine outlet gas temperature: (See NOTE 7)

Take-off (5 minutes maximum)	730°C
Maximum continuous	705°C
Maximum for acceleration	705°C
Starting maximum gas temperature	800°C

Maximum permissible oil inlet temperature:

Continuous operation (See NOTE 13)	125°C

Maximum permissible air bleed extraction of primary engine airflow 7.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating)
from sea level to 27,800 feet 285 knots

M_{MO} (Maximum operating)
27,800 ft. and above 0.750 M

V_A (Maneuvering)

Sea level	181 knots
10,000 ft.	182 knots
20,000 ft.	183 knots
30,000 ft.	195 knots
40,000 ft.	212 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15°	210 knots
25°	160 knots
50° or 45° (See NOTE 23)	145 knots

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	93 knots
V_{MCG} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	84 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage lower skin immediately aft of the equipment bay access hatch.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps retracted)

	Fwd of Datum				Aft of Datum			
	In-Flight		T.O. & Land		Autopilot disengaged		Autopilot engaged	
Wt. Lbs.	% SMC	In.	% SMC	In.	% SMC	In.	% SMC	In.
21,200	22.40	0.55	23.60	(0.54)*	33.80	9.80	32.60	8.70
20,550	-	-	-	-	34.30	10.20	33.00	9.00
19,000	-	-	-	-	34.10	9.95	32.80	8.82
17,800	18.80	3.78	20.20	2.52	-	-	-	-
16,800	-	-	-	-	33.10	9.10	-	-
16,600	-	-	-	-	-	-	31.60	7.80
13,000	18.00	4.51	20.00	2.71	-	-	-	-
12,350	-	-	-	-	37.50	13.10	35.50	11.30
12,100	18.00	4.51	20.00	2.71		-	-	-
10,800	-	-	26.00	(2.71)*	37.50	13.10	35.50	11.30

*(Aft of Datum)
Straight line variation between weights

Item (Extending)	Moment Change In. lbs.
Wing Flaps 15^{o}	+ 538
25^{o}	+ 879
50^{o} or 45^{o} (See NOTE 23)	+1,593
Main Landing Gear	-1,800
Nose Landing Gear	+1,380

The airplane is normally weighed with wing flaps retracted

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum Ramp Weight	21,200 lbs. (See NOTE 12)
Maximum Brake release weight	21,200 lbs.
Maximum Landing Weight	19,550 lbs.
Maximum Zero Fuel Weight	13,000 lbs. (See NOTE 12)

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Total	1,231.5	8,210	5.79

Oil Capacity — Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In.Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — Same as listed previously for Hawker Siddeley Models DH.125 Series 1A and the HS.125 Series 1B

III. Hawker Siddeley Model DH.125 Series 3A (Transport Aircraft), Approved November 7, 1966.
Hawker Siddeley Model HS.125 Series 3B (Transport Aircraft), Approved May 28, 1999. (See NOTE 52)
(The DH.125 Series 3A aircraft and the HS.125 Series 3B aircraft differs respectively from the DH.125 Series 1A-522 aircraft and the HS.125 Series 1B-522 aircraft in the following major features: (i) increased maximum ramp, brake release, landing and zero fuel weights. (ii) increased Mmo. (iii) Vmo - 285 knots reducing linearly to 273 knots between 27,200 feet and 30,800 feet).

Engines — 2 Bristol Siddeley Viper 522 turbine engines.

Fuel — Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22f Type 2, ASTM D.1655 Jet B. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
Maximum permissible engine rotor Operating speed (5 minutes maximum)	100% (13,760 r.p.m.)
Maximum permissible turbine outlet gas temperature: (See NOTE 7)	
Take-off (5 minutes maximum)	730°C
Maximum continuous	705°C
Maximum for acceleration	705°C
Starting maximum gas temperature	800°C
Maximum permissible oil inlet temperature: Continuous operation (See NOTE 13)	125°C
Maximum permissible air bleed extraction of primary engine airflow	7.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating) from sea level to 27,200 feet and decreasing linearly to 273 knots at 30,800 feet	285 knots
M_{MO} (Maximum operating) 30,800 ft. and above (See NOTE 15)	0.765 M

Airspeed Limits (IAS)(continued)

V_A (Maneuvering)

Sea level	185 knots
10,000 ft.	185 knots
20,000 ft.	185 knots
30,000 ft.	195 knots
40,000 ft.	210 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15^o	210 knots
25^o	160 knots
50^o or 45^o (See NOTE 23)	145 knots

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	93 knots
V_{MCG} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	84 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage lower skin immediately aft of the equipment bay access hatch.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps retracted)

	Fwd of Datum				Aft of Datum			
	In-Flight		T.O. & Land		Autopilot disengaged		Autopilot engaged	
Wt. Lbs.	% SMC	In.	% SMC	In.	% SMC	In.	% SMC	In.
21,700	23.64	(0.57)*	24.87	(1.68)*	33.67	9.62	32.44	8.51
21,200	22.40	0.55	23.60	(0.54)*	-	-	-	-
20,550	-	-	-	-	34.30	10.20	33.00	9.00
19,000	-	-	-	-	34.10	9.95	32.80	8.82
17,800	18.80	3.78	20.20	2.52	-	-	-	-
16,800	-	-	-	-	33.10	9.10	-	-
16,600	-	-	-	-	-	-	31.60	7.80
13,000	18.00	4.51	20.00	2.71	-	-	-	-
12,350	-	-	-	-	37.50	13.10	35.50	11.30
12,100	18.00	4.51	20.00	2.71		-	-	-
10,800	-	-	26.00	(2.71)*	37.50	13.10	35.50	11.30

*(Aft of Datum)

Straight line variation between weights

Item (Extending)	Moment Change In.Lbs.
Wing Flaps 15^o	+ 538
25^o	+ 879
50^o or 45^o (See NOTE 23)	+1,593
Main Landing Gear	-1,800
Nose Landing Gear	+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means: Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum Ramp Weight	21,700 lbs. (See NOTE 12)
Maximum Brake Release Weight	21,700 lbs.
Maximum Landing Weight	20,000 lbs.
Maximum Zero Fuel Weight	13,500 lbs. (See NOTE 12)

Minimum Crew: For all flights, 2 pilots

Maximum Passengers: 8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and Lines	1.5	10	81.00
Total	1,231.5	8,210	5.79

Oil Capacity: Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal.	Maximum Weight Lb.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude: 40,000 feet (See NOTE 9)

Serial Numbers Eligible: 25015, 25062, 25069, 25111 through 25172

IV. Hawker Siddeley Model DH.125 Series 1A/R-522 (Transport Aircraft), Approved August 9, 1967.
Hawker Siddeley Model HS.125 Series 1B/R-522 (Transport Aircraft), Approved May 28, 1999. (See NOTE 52)
(The DH.125 Series 1A/R-522 aircraft and the HS.125 Series 1B/R-522 aircraft differs respectively from the DH.125 Series 1A-522 aircraft and the HS.125 Series 1B-522 by the incorporation of Modifications No. 251700 and 255640, long-range fuel tank, modified flaps and main landing gear doors.) (See NOTE 10).

Engines: 2 Bristol Siddeley Viper 522 turbine engines.

Fuel: Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22f Type 2, ASTM D.1655 Jet B. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
Maximum permissible engine rotor	100%
Operating speed (5 minutes maximum)	(13,760 r.p.m.)

Maximum permissible turbine outlet gas temperature: (See NOTE 7)

Take-off (5 minutes maximum)	730°C
Maximum continuous	705°C
Maximum for acceleration	705°C
Starting maximum gas temperature	800°C

Maximum permissible oil inlet temperature: Continuous operation (See NOTE 13)	125°C
Maximum permissible air bleed extraction of primary engine airflow	7.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating)

from sea level to 27,800 feet with fuel in long range tank	260 knots
from sea level to 27,800 feet with long range tank empty	285 knots

M_{MO} (Maximum operating)

27,800 ft. and above	0.750 M

V_A (Maneuvering)

Sea level	189 knots
10,000 ft.	190 knots
20,000 ft.	196 knots
30,000 ft.	202 knots
35,000 ft.	207 knots
40,000 ft.	201 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15°	210 knots
25°	160 knots
50° or 45° (See NOTE 23)	145 knots

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	93 knots
V_{MCG} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	84 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage lower skin immediately aft of the equipment bay access hatch.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps retracted)

Wt. lbs.	Fwd. of Datum % SMC	Fwd. of Datum In.	Aft of Datum % SMC	Aft of Datum In.
22,200	29.53	(5.89)*	37.87	13.41
22,100	-	-	38.00	13.53
21,400	25.53	(2.28)*	-	-
19,200	-	-	37.93	13.47
18,840	-	-	34.00	9.92
17,750	22.60	0.37	-	-
17,400	-	-	33.47	9.44
13,200	23.00	0.00	-	-
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

Item (Extending)	Moment Change In. Lbs.
Wing Flaps 15^{o}	+ 538
25^{o}	+ 879
50^{o} or 45^{o} (See NOTE 23)	+1,593
Main Landing Gear	-1,800
Nose Landing Gear	+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum Ramp Weight	22,300 lbs.
Maximum Brake Release Weight	22,200 lbs.
Maximum Landing Weight	19,550 lbs.
Maximum Zero Fuel Weight	13,200 lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Long Range Tank	134.5	896	88.70
Total	1,366.0	9,106	13.95

Oil Capacity — Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — Same as listed previously for Hawker Siddeley Models DH.125 Series 1A and the HS.125 Series 1B

V. Hawker Siddeley Model DH.125 Series 3A/R (Transport Aircraft), Approved August 9, 1967.
Hawker Siddeley Model HS.125 Series 3B/R (Transport Aircraft), Approved May 28, 1999. (See NOTE 52)
(The DH.125 Series 3A/R aircraft and the HS.125 Series 3B/R aircraft differs respectively from the DH.125 Series 3A aircraft and the HS.125 Series 3B by the incorporation of Modifications No. 251700 and 255640, long-range fuel tank, modified flaps and main landing gear doors). (See NOTE 10)

Engines — 2 Bristol Siddeley Viper 522 turbine engines.

Fuel — Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22f Type 2, ASTM D.1655 Jet B. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
Maximum permissible engine rotor	100%
Operating speed (5 minutes maximum)	(13,760 r.p.m.)
Maximum permissible turbine outlet gas temperature: (See NOTE 7)	
Take-off (5 minutes maximum)	730°C
Maximum continuous	705°C
Maximum for acceleration	705°C
Starting maximum gas temperature	800°C
Maximum permissible oil inlet temperature: Continuous operation (See NOTE 13)	125°C
Maximum permissible air bleed extraction of primary engine airflow	7.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating)	
from sea level to 30,800 feet with fuel in long range tank	260 knots
from sea level to 27,200 feet with long range tank empty decreasing linearly to 273 knots at 30,800 feet.	285 knots

Airspeed Limits (IAS)(Cont.)

M_{MO} (Maximum operating)
30,800 ft. and above 0.765 M

V_A (Maneuvering)

Sea level	190 knots
10,000 ft.	191 knots
20,000 ft.	197 knots
30,000 ft.	203 knots
35,000 ft.	208 knots
40,000 ft.	201 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15^o	210 knots
25^o	160 knots
50^o or 45^o (See NOTE 23)	145 knots

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	93 knots
V_{MCG} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	84 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
22,700	30.53	(6.79)*	37.20	12.81
22,150	-	-	38.00	13.53
21,700	25.80	(2.52)*	-	-
19,200	-	-	37.93	13.47
18,850	-	-	34.00	9.92
17,750	22.47	0.48	-	-
17,400	-	-	33.47	9.44
13,000	23.00	0.00	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

Item (Extending)		Moment Change In. Lbs.
Wing Flaps	15^o	+ 538
	25^o	+ 879
	50^o or 45^o (See NOTE 23)	+1,593
Main Landing Gear		-1,800
Nose Landing Gear		+1,380

C.G. Range (Gear and Flaps Retracted)(Cont.)

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum Ramp Weight	22,800 lbs.
Maximum Brake Release Weight	22,700 lbs.
Maximum Landing Weight	20,000 lbs.
Maximum Zero Fuel Weight	13,700 lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Long Range Tank	134.5	896	88.70
Total	1,366.0	9,106	13.95

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lb.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible

Same as listed previously for Hawker Siddeley Models DH.125 Series 3A and the HS.125 Series 3B

VI. Hawker Siddeley Model DH.125 Series 1A/S-522 (Transport Aircraft), Approved February 15, 1968
Hawker Siddeley Model HS.125 Series 1B/S-522 (Transport Aircraft), Approved May 28, 1999. (See NOTE 52)
(The DH.125 Series 1A/S-522 aircraft and the HS.125 Series 1B/S-522 aircraft differs respectively from the DH.125 Series 1A-522 aircraft and the HS.125 Series 1B-522 aircraft by the incorporation of Modification No. 251867 which introduces structural additions enabling the aircraft to be operated to the same limitations as the DH.125 Series 3A or the HS.125 Series 3B aircraft respectively except for the maximum landing weight which remains at 19,550 lbs., and maximum operating altitude). (See NOTE 11).

Engines	2 Bristol Siddeley Viper 522 turbine engines.	
Fuel	Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22f Type 2, ASTM D.1655 Jet B. (See NOTE 4).	
Engine Limits	Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
	Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
	Maximum permissible engine rotor	100%
	Operating speed (5 minutes maximum)	(13,760 r.p.m.)
	Maximum permissible turbine outlet gas temperature: (See NOTE 7)	
	Take-off (5 minutes maximum)	730°C
	Maximum continuous	705°C
	Maximum for acceleration	705°C
	Starting maximum gas temperature	800°C
	Maximum permissible oil inlet temperature:	
	Continuous operation (See NOTE 13)	125°C
	Maximum permissible air bleed extraction of primary engine airflow	7.5%
Airspeed Limits (IAS)	V_{MO} (Maximum operating) from sea level to 27,200 feet and decreasing linearly to 273 knots at 30,800 feet	285 knots
	M_{MO} (Maximum operating) 30,800 ft. and above (See NOTE 15)	0.765 M
	V_A (Maneuvering)	
	Sea level	185 knots
	10,000 ft.	185 knots
	20,000 ft.	185 knots
	30,000 ft.	195 knots
	40,000 ft.	210 knots
	Straight line variation between points shown.	
	V_{FE} (Flap speeds)	
	Deflection	
	15°	210 knots
	25°	160 knots
	50° or 45° (See NOTE 23)	145 knots
	V_{LO} (Landing gear operation)	
	Retract	210 knots
	Extend	210 knots
	V_{LE} (Landing gear extended)	210 knots
	V_{MC} (Minimum control speed)	
	V_{MCA} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	93 knots

Airspeed Limits (IAS)(Cont.)

V_{MCG} (with flaps at 0^o or 15^o at sea level for temperatures below 10^oC)	84 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage lower skin immediately aft of the equipment bay access hatch.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps retracted)

	Fwd of Datum				Aft of Datum			
	In-Flight		T.O. & Land		Autopilot disengaged		Autopilot engaged	
Wt. Lbs.	% SMC	In.	% SMC	In.	% SMC	In.	% SMC	In.
21,700	23.64	(0.57)*	24.87	(1.68)*	33.67	9.62	32.44	8.51
21,200	22.40	0.55	23.60	(0.54)*	-	-	-	-
20,550	-	-	-	-	34.30	10.20	33.00	9.00
19,000	-	-	-	-	34.10	9.95	32.80	8.82
17,800	18.80	3.78	20.20	2.52	-	-	-	-
16,800	-	-	-	-	33.10	9.10	-	-
16,600	-	-	-	-	-	-	31.60	7.80
13,000	18.00	4.51	20.00	2.71	-	-	-	-
12,350	-	-	-	-	37.50	13.10	35.50	11.30
12,100	18.00	4.51	20.00	2.71		-	-	-
10,800	-	-	26.00	(2.71)*	37.50	13.10	35.50	11.30

*(Aft of Datum)
Straight line variation between weights

Item (Extending)		Moment Change In. Lbs.
Wing Flaps	15^o	+ 538
	25^o	+ 879
	50^o or 45^o (See NOTE 23)	+1,593
Main Landing Gear		-1,800
Nose Landing Gear		+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum Ramp Weight	21,700 lbs. (See NOTE 12)
Maximum Brake Release Weight	21,700 lbs.
Maximum Landing Weight	19,550 lbs.
Maximum Zero Fuel Weight	13,500 lbs. (See NOTE 12)

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm. In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Total	1,231.5	8,210	5.79

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — Same as listed previously for Hawker Siddeley Models DH.125 Series 1A and the HS.125 Series 1B

VII. Hawker Siddeley Model DH.125 Series 3A/RA (Transport Aircraft), Approved February 15, 1968 (See NOTE 46)
Hawker Siddeley Model HS.125 Series 3B/RA (Transport Aircraft), Approved May 28,1999. (See NOTES 46 & 52).

The DH.125 Series 3A/RA aircraft and the HS.125 Series 3B/RA aircraft differs respectively from the DH.125 Series 3A/R aircraft and the HS.125 Series 3B/R by (i) incorporation of Modification No. 251916 which introduces structural additions to permit a maximum zero fuel weight of 14,200 lbs., (ii) a maximum ramp weight of 23,100 lbs. (See NOTE 11)

Hawker Siddeley Model HS.125 Series 3B/RB (Transport Aircraft), Approved May 28, 1999. (See NOTE 52)

The DH.125 Series 3B/RB aircraft differs respectively from the DH.125 Series 3B/RA aircraft by incorporation of Modification No. 252024 which increases the maximum ramp weight and the maximum take off weight.

Hawker Siddeley Model HS.125 Series 3B/RC (Transport Aircraft), Approved May 28, 1999. (See NOTE 52)

The DH.125 Series 3B/RC aircraft differs respectively from the DH.125 Series 3B/RA aircraft by incorporation of modifications to enable it to be used for checking navigational aids by (i) installation of a special four seat cabin configuration and (ii) installation of special Avionics and Flight Inspection equipment.

Engines — 2 Bristol Siddeley Viper 522 turbine engines.

Fuel — Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22f Type 2, ASTM D.1655 Jet B. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
Maximum permissible engine rotor	100%
Operating speed (5 minutes maximum)	(13,760 r.p.m.)

Maximum permissible turbine outlet gas temperature: (See NOTE 7)

Take-off (5 minutes maximum)	730°C
Maximum continuous	705°C
Maximum for acceleration	705°C
Starting maximum gas temperature	800°C

Maximum permissible oil inlet temperature: Continuous operation (See NOTE 13)	125°C
Maximum permissible air bleed extraction of primary engine airflow	7.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating)

from sea level to 30,800 feet with fuel in long range tank	260 knots
from sea level to 27,200 feet with long range tank empty decreasing linearly to 273 knots at 30,800 feet.	285 knots

V_{MO} (Maximum operating) (with Mod. 25A767A)

from sea level to 30,800 feet with fuel in long range tank	255 knots
from sea level to 27,500 feet with long range tank empty decreasing linearly to 265 knots at 31,980 feet.	281 knots

M_{MO} (Maximum operating)

30,800 ft. and above	0.765 M

M_{MO} (Maximum operating) (3B/RB only)

30,800 ft. and above	0.755 M

V_A (Maneuvering)

Sea level	190 knots
10,000 ft.	191 knots
20,000 ft.	197 knots
30,000 ft.	203 knots
35,000 ft.	208 knots
40,000 ft.	201 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15°	210 knots
25°	160 knots
50° or 45° (See NOTE 23)	145 knots

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	93 knots
V_{MCG} (with flaps at 0° or 15° at sea level for temperatures below 10°C)	84 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
22,700	27.20	(3.78)*	38.00	13.53
22,200	25.20	(1.98)*	-	-
19,200	-	-	38.00	13.53
18,950	22.47	0.48	-	-
18,850	-	-	34.00	9.92
18,400	-	-	33.80	9.74
14,200	23.00	0.00	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (for 3B/RB only) (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,300	29.67	(6.01)*	37.07	12.69
22,400	28.73	(5.17)*	-	-
22,400	26.00	(2.70)*	-	-
22,100	-	-	38.00	13.53
19,200	-	-	38.00	13.53
18,850	-	-	34.00	9.92
18,400	22.47	0.48	-	-
17,450	-	-	33.33	9.32
16,800	22.40	0.55	-	-
14,200	-	-	36.13	9.57
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

Item (Extending)		Moment Change In.Lbs.
Wing Flaps	15^{o}	+538
	25^{o}	+879
	50^{o} or 45^{o} (See NOTE 23)	+1,593
Main Landing Gear		-1,800
Nose Landing Gear		+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		(3B/RB only)
Maximum Ramp Weight	23,100 lbs.	23,300 Lbs.
Maximum Brake-Release Weight	22,700 lbs.	23,300 Lbs.
Maximum Landing Weight	20,000 lbs.	20,000 Lbs.
Maximum Zero Fuel Weight	14,200 lbs.*	14,200 Lbs.

* (See NOTE 46)

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Ventral tank	134.5	896	88.70
Total	1,366.0	9,106	13.95

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — Same as listed previously for Hawker Siddeley Models DH.125 Series 3A/R and HS.125 Series 3B/R. (See NOTE 61)

VIII. Hawker Siddeley Model DH.125 Series 400A (Transport Aircraft), Approved November 15, 1968 (See NOTE 16)
Hawker Siddeley Model HS.125 Series 400B (Transport Aircraft), Approved May 28, 1999. (See NOTES 16 & 53).
(The DH.125 Series 400A aircraft and the HS.125 Series 400B aircraft differs respectively from the DH.125 Series 3A/RA aircraft and the HS.125 Series 3B/RA aircraft in the following major features: (i) increased maximum ramp and brake-release weights; (ii) introduction of an outward-opening main entry door)

Beechcraft Hawker Model BH.125 Series 400A (Transport Aircraft) Approved 14 July 1970 (See NOTE 16)
(The Hawker Siddeley Model DH.125 Series 400A is, from aircraft Serial Number 25230 and subsequent, identified as the Beechcraft Hawker Model BH.125 Series 400A. The BH.125 Series 400A is, in all respects, identical to the DH.125 Series 400A except that the aircraft data plate, the control column central motif and the external nameplate on the fuselage nose have been altered to incorporate the revised identification.)

Hawker Siddeley Model HS.125 Series 400B/1 (Transport Aircraft), Approved May 28, 1999. (See NOTES 16 & 53).
(The HS.125 Series 400B/1 aircraft were originally manufactured as a HS.125 Series 400A aircraft converted to a HS.125 Series 400B aircraft and then reconverted to the equivalent of a HS.125 Series 400A aircraft.)

Hawker Siddeley Model HS.125 Series 401B (Transport Aircraft), Approved May 28, 1999. (See NOTE 53)
(The HS.125 Series 401B aircraft differs respectively from the HS.125 Series 400B aircraft in the following major features: (i) increased maximum take off weight and zero fuel weight and (ii) cabin loading altered (See NOTE 8).

Hawker Siddeley Model HS.125 Series 403B (Transport Aircraft), Approved May 28, 1999. (See NOTE 53)
Hawker Siddeley Model HS.125 Series 403A(C) (Transport Aircraft), Approved May 28, 1999. (See NOTE 53)
(The HS.125 Series 403B and the HS.125 Series 403A(C) aircraft differs respectively from the HS.125 Series 400A aircraft in the following major features: (i) increased maximum take off weight, zero fuel weight, and ramp weight and (ii) cabin loading was altered (See NOTE 8). The HS.125 Series 403A(C) aircraft was for Canadian registry.

The following details are applicable to both the Hawker Siddeley Models DH.125 Series 400A, HS.125 Series 400B, Beechcraft Hawker Model BH.125 Series 400A. All other models are the same except as noted. (See NOTE 35)

Engines

2 Bristol Siddeley Viper 522 turbine engines.

Fuel

Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22 Type 2, ASTM D.1655 Jet B, Mil-t-5624 JP4 Grade. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,330
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,100
Maximum permissible engine rotor operating speed (5 minutes maximum)	100% (13,760 r.p.m.)

Maximum permissible turbine outlet gas temperature: (See NOTE 7)

Take-off (5 minutes maximum)	730°C
Maximum continuous	705°C
Maximum for acceleration	705°C
Starting maximum gas temperature	800°C

Maximum permissible oil inlet temperature: Continuous operation (See NOTE 13)	125°C
Maximum permissible air bleed extraction of primary engine airflow	7.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating)

from sea level to 30,800 feet with fuel in long range tank	260 knots
from sea level to 27,200 feet with long range tank empty decreasing linearly to 273 knots at 30,800 feet.	285 knots

V_{MO} (Maximum operating) (with Mod. 252243C or 256403D, Series 401B, Series 403A(C) or Series 403B)

from sea level to 30,800 feet with fuel in long range tank	257 knots
from sea level to 27,500 feet with long range tank empty decreasing linearly to 270 knots at 31,350 feet. (See NOTE 16)	282 knots

Airspeed Limits (IAS)(Cont.)

M_{MO} (maximum operating) (See NOTE 15) 30,800 ft. and above	0.765 M
M_{MO} (maximum operating) (Series 401B and Series 403B) 30,800 ft. and above	0.755 M

V_A (Maneuvering)

Sea level	193 knots
10,000 feet	195 knots
20,000 feet	201 knots
30,000 feet	208 knots
35,000 feet	213 knots
40,000 feet	209 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15^{o}	210 knots
25^{o}	160 knots
50^{o} or 45^{o} (See NOTE 23)	145 knots

V_{LO} (Landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (Landing gear extended) 210 knots

V_{MC} (Minimum control speed)

V_{MCA} (with flaps at 0^{o} or 15^{o} at sea level for temperatures below 10^{o}C)	93 knots
V_{MCG} (with flaps at 0^{o} or 15^{o} at sea level for temperatures below 10^{o}C)	84 knots

Datum — The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC) — 90.24 in. The leading edge of the SMC is 20.76 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,300	29.67	(6.01)*	37.13	12.75
22,400	28.73	(5.17)*	-	-
22,400	26.00	(2.70)*	-	-
22,100	-	-	38.00	13.53
19,200	-	-	38.00	13.53
18,850	-	-	34.00	9.92
18,400	22.45	.50	-	-
17,400	-	-	33.33	9.32
16,800	22.40	.55	-	-
14,200	-	-	36.00	11.73
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (with Mod. 252243C)(See NOTE 16) (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,300	29.40	(5.77)*	37.13	12.75
22,900	-	-	38.00	13.53
22,700	29.13	(5.53)*	-	-
22,700	25.93	(2.64)*	-	-
19,200	-	-	38.00	13.53
18,850	-	-	34.00	9.92
18,400	22.45	0.50	-	-
17,400	-	-	33.33	9.32
16,800	22.40	0.55	-	-
14,200	-	-	36.00	11.73
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (with Mod. 256403D)(See NOTE 16)

(Gear and Flaps Retracted)	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,300	29.40	(5.77)*	37.00	12.63
22,900	29.20	(5.59)*	-	-
22,900	26.07	(2.77)*	-	-
21,900	-	-	38.00	13.53
19,200	-	-	38.00	13.53
18,900	-	-	34.00	9.92
18,500	22.45	0.50	-	-
17,400	-	-	33.33	9.32
16,200	22.40	0.55	-	-
14,200	-	-	36.00	11.73
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (Series 401B)

(Gear and Flaps Retracted)	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,600	29.73	(6.07)*	34.20	10.10
23,300	-	-	37.00	12.63
22,900	29.20	(5.59)*	-	-
22,900	26.07	(2.77)*	-	-
21,900	-	-	38.00	13.53
19,200	-	-	38.00	13.53
18,900	-	-	33.93	9.86
18,500	22.45	0.50	-	-
17,400	-	-	33.33	9.32
16,200	22.40	0.55	-	-
14,200	-	-	36.00	11.73
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (Series 403A(C) and Series 403B)

(Gear and Flaps Retracted)	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,600	29.53	(5.89)*	34.20	10.10
23,300	-	-	35.13	10.94
23,300	-	-	37.00	12.63
22,900	29.20	(5.59)*	-	-
22,900	26.07	(2.77)*	-	-
21,900	-	-	38.00	13.53
19,200	-	-	38.00	13.53
18,900	-	-	33.93	9.86
18,500	22.45	0.50	-	-
17,400	-	-	33.33	9.32
16,200	22.40	0.55	-	-
14,200	-	-	36.00	11.73
13,000	-	-	37.53	13.11
12,000	23.00	0.00	-	-
11,000	28.00	(4.51)*	37.53	13.11

*(Aft of Datum)
Straight line variation between weights.

Item (extending)		Moment Change In.Lbs
Wing flaps	15^{o}	+538
	25^{o}	+879
	50^{o} or 45^{o} (See NOTE 23)	+1,593
Main Landing Gear		-1,800
Nose Landing Gear		+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

	(with Mod. 252243C)	(with Mod. 256403D)
Maximum Ramp Weight	23,600 Lbs.	23,800 Lbs.
Maximum Brake-Release Weight	23,300 Lbs.	23,300 Lbs.
Maximum Landing Weight	20,000 Lbs.	20,000 Lbs.
Maximum Zero Fuel Weight	14,500 Lbs.	14,700 Lbs.

	(Series 401B)	(Series 403A(C) & Series 403B)
Maximum Ramp Weight	23,600 lbs.	23,800 lbs.
Maximum Brake-Release Weight	23,600 lbs.	23,600 lbs.
Maximum Landing Weight	20,000 lbs.	20,000 lbs.
Maximum Zero Fuel Weight	14,500 lbs.	14,700 lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205 to 260	60	210
8 seater	205 to 250	60	160
Forward cabin			
(a) Side floor	260 to 303.85	50	
(b) Center floor	260 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395	50	
(b) Center floor	303.85 to 395	60	
Aft	395 to 425	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Long Range Tank	134.5	896	88.70
Total	1,366.0	9,106	13.95

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.87	14	82	1153
No. 2	1.87	14	82	1153
Total	3.74	28	82	2306

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — DH.125 Series 400A and HS.125 Series 400B: 25173 through 25229 (See NOTE 61)
BH.125 Series 400A: 25230 through 25290 (See NOTE 61)

IX. Beechcraft Hawker Model BH.125 Series 600A (Transport Aircraft) Approved August 17, 1972 (See NOTE 17).
Hawker Siddeley Model HS.125 Series 600B (Transport Aircraft), Approved May 28, 1999. (See NOTES 17 & 54).
The BH.125 Series 600A aircraft and the HS.125 Series 600B differs respectively from the BH.125 Series 400A and the HS.125 Series 400B in the following major features: (i) Introduction of Rolls Royce (1971) Ltd., Bristol Engine Division Viper 601-22 engines, (ii) increased maximum ramp, brake release, landing and zero fuel weights, (iii) increased maximum operating speed (V_{MO}), rough air speed (V_{RA}), flap operating speeds (V_{FE}), and landing gear operating speed (V_{LE}), (iv) increased fuselage length, (v) increased fuel capacity by the addition of an extra fuel tank in the dorsal fairing, (vi) revised aileron tab arrangement and aileron control gearing, (vii) aerodynamic improvements providing better aircraft aerodynamic lines.

Hawker Siddeley Model HS.125 Series 600A (Transport Aircraft) Approved January 6, 1976 (See NOTE 17).
The Beechcraft Hawker Model BH.125 Series 600A, is, from aircraft Serial No. 256055, identified as the Hawker Siddeley Model HS.125 Series 600A. The Hawker Siddeley Model HS.125 Series 600A is in all respects identical to the Beechcraft Hawker Model BH.125 Series 600A except that the aircraft data plate, the control column central motif and the external nameplate on the fuselage nose have all been altered to reflect the revised identification.

Hawker Siddeley Model HS.125 Series 600B/1 (Transport Aircraft), Approved May 28, 1999. (See NOTES 17, 54 and 55).
Hawker Siddeley Model HS.125 Series 600B/2 (Transport Aircraft), Approved May 28, 1999. (See NOTES 17, 54 and 55).
Hawker Siddeley Model HS.125 Series 600B/3 (Transport Aircraft), Approved May 28, 1999. (See NOTES 17, 54 and 55).
The following details are applicable to both the Beechcraft Hawker Model BH.125 Series 600A and the Hawker Siddeley Models HS.125 Series 600A and Series 600B, 600B/1, 600B/2 and 600B/3 aircraft. (See NOTE 18).

Engines — 2 Rolls Royce (1971) Ltd., Bristol Engine Division Viper 601-22 turbine engines

Fuel — Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23 Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22 Type 2, ASTM D.1655 Jet B, Mil-T-5624 JP4. (See NOTE 4).

Engine Limits

Take-off static thrust, standard day, sea level conditions (5 minutes maximum) lbs.	3,675
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,675
Maximum permissible engine rotor operating speed	100%(13,760 r.p.m.)
Maximum permissible turbine outlet gas temperature:	
Take-off (5 minutes maximum)	725°C
Maximum continuous	715°C
Maximum for acceleration	715°C
Starting maximum gas temperature	700°C
Maximum permissible oil inlet temperature:	
Continuous operation	145°C
Maximum permissible air bleed extraction of primary engine airflow	5.5%

Airspeed Limits (IAS)

V_{MO} (Maximum operating)
when the dorsal and/or ventral fuel tank contains fuel 280 knots
when the dorsal and the ventral fuel tanks are empty 300 knots

V_{MO} (Maximum operating) (with Mod. 252320)(See NOTE 17)
when the dorsal and/or ventral fuel tank contains fuel 280 knots
from sea level to 12,400 feet with dorsal and ventral tanks empty 320 knots
decreasing linearly to 292 knots at 29,200 feet.

M_{MO} (maximum operating)
30,800 ft. and above 0.755 M

M_{MO} (maximum operating) (with Mod. 252320)(See NOTE 17)
29,200 ft. and above 0.78 M

V_A (Maneuvering)

Sea level	190 knots
10,000 feet	193 knots
20,000 feet	196 knots
30,000 feet	201 knots
35,000 feet	205 knots
40,000 feet	212 knots

Straight line variation between points shown.

Airspeed Limits (IAS)(Cont.)

V_{FE} (Flap speeds)

Deflection	
15^o	220 knots
25^o	175 knots
45^o	160 knots

V_{LO} (Landing gear operation)

Retract	220 knots
Extend	220 knots

V_{LE} (Landing gear extended) 220 knots

V_{MC} (Minimum control speed)
V_{MCA} (with flaps at 0^o or 15^o at sea level
for temperatures below 10^oC) 96 knots
V_{MCG} (with flaps at 0^o or 15^o at sea level
for temperatures below 10^oC) 90 knots

Datum. The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC) 90.24 in. The leading edge of the SMC is 20.76 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
25,000	26.40	(3.06)*	34.73	10.58
24,850	-	-	35.00	10.82
24,590	26.13	(2.82)*	-	-
24,590	24.27	(1.14)*	-	-
23,700	23.67	(0.60)*	-	-
23,700	20.53	2.23	-	-
20,800	-	-	35.00	10.82
20,510	-	-	30.40	6.67
20,300	17.07	5.36	-	-
19,000	-	-	29.53	5.89
15,550	16.00	6.32	-	-
14,700	-	-	32.00	8.12
12,800	16.00	6.32	31.53	7.69

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (with Mod. 252320) (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
25,500	26.87	(3.49)*	33.80	9.74
25,100	26.47	(3.13)*	-	-
25,100	24.53	(1.38)*	-	-
24,850	-	-	35.00	10.82
24,200	23.93	(0.83)*	-	-
24,200	21.00	1.81	-	-
20,800	-	-	35.00	10.82
20,510	-	-	30.40	6.67
20,300	17.07	5.36	-	-
19,000	-	-	29.53	5.89
15,550	16.00	6.32	-	-
14,700	-	-	32.00	8.12
12,800	16.00	6.32	31.53	7.69

*(Aft of Datum)
Straight line variation between weights.

C.G. Range (with Mod. 256663) (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
25,500	26.60	(3.49)*	32.00	9.74
25,100	26.40	(3.13)*	-	-
25,100	24.53	(1.38)*	-	-
24,200	23.93	(0.83)*	-	-
24,200	21.00	1.81	-	-
20,650	-	-	32.00	10.82
20,500	-	-	30.33	10.82
20,300	17.07	5.36	-	-
19,150	-	-	29.47	10.82
15,550	16.00	6.32	-	-
14,700	-	-	32.00	8.12
12,800	16.00	6.32	31.40	7.58

*(Aft of Datum)
Straight line variation between weights.

Item (Extending)	Moment Change In.Lbs.
Wing Flaps 15^{o}	+538
25^{o}	+879
45^{o}	+1,593
Main landing gear	-1,980
Nose landing gear	+1,380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		(With Mod 252320)
Maximum Ramp Weight	25,000 lbs.	25,500 lbs.
Maximum Brake-Release Weight	25,000 lbs.	25,500 lbs.
Maximum Landing Weight	22,000 lbs.	22,000 lbs.
Maximum Zero Fuel Weight	15,550 lbs.	16,050 lbs.

	(With Mod. 256663)
Maximum Ramp Weight	25,500 lbs.
Maximum Brake-Release Weight	25,500 lbs.
Maximum Landing Weight	22,000 lbs.
Maximum Zero Fuel Weight	16,050 lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

15

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward	180.25 to 223.11	100	310
Forward cabin			
(a) Side floor	245.85 to 303.85	50	
b) Center floor	245.85 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	397.8 to 422.3	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	611.0	4,070	5.60
Tank 2	611.0	4,070	5.60
Engines and lines	1.5	10	81.00
Long Range (ventral tank)	134.5	896	88.60
Dorsal tank	61.0	406	119.30
Total	1,419.0	9,452	18.43

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	2.03	15.3	83	1224
No. 2	2.03	15.3	83	1224
Total	4.06	30.6	83	2448

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible

BH.125 Series 600A and HS.125 Series 600B: 25256, 25258, 256001 through 256035, and 256037 through 256054
HS.125 Series 600A: 256055 through 256071

X. Hawker Siddeley Model HS.125 Series 700A (Transport Aircraft), Approved May 20, 1977
Hawker Siddeley Model HS.125 Series 700B (Transport Aircraft), Approved May 28, 1999. (See NOTE 56)
The HS.125 Series 700A and HS.125 Series 700B aircraft differs respectively from the BH/HS.125 Series 600A and the HS.125 Series 600B aircraft in the following major respects: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 601-22 engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance, (iv) Provisions for a New Automatic Flight Control System - Collins FCS.80, (v) Addition of a single point pressure refuel/defuel system, (vi) Reduction in certificated taxiing and take-off (brake release) weights, (vii) Reduction of M_{MO} from 0.78 (Post Modification 252320 Part A) to 0.77.

The following details are applicable to both the Hawker Siddeley Model HS.125 Series 700A and Series 700B aircraft. (See NOTES 20, 24 & 27)

Engines

2 Garrett AiResearch TFE 731-3 turbofan engines, or
2 Garrett AiResearch TFE 731-3R turbofan engines (See NOTE 20)

Fuel

Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23h Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22 Type 2, ASTM D.1655 Jet B, Mil-T-5624 JP4. (See NOTE 28).

Engine Limits

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	3,700	3,880
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,700	3,700
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	101.5 (21,000 rpm)	101.5 (21,000 rpm)
H.P. Shaft (N2)	100 (29,692 rpm)	100 (29,989 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	907°C	929°C
Take-off (10 minutes maximum)	917°C	939°C
Take-off (instantaneous)	927°C	949°C
Maximum continuous	885°C	885°C
Engine starting and relighting (unrestricted)	907°C	907°C
Engine starting and relighting (10 seconds)	927°C	927°C
Engine starting and relighting (5 seconds)	above 927°C	above 927°C
Maximum permissible oil temperature:		
Sea level to 30,000 ft.	127°C	127°C
Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
Minimum permissible oil temperature:		
Engine starting	-40°C	-40°C
Before take-off	+30°C	+30°C

	Maximum permissible air bleed extraction:		
	L.P. air source	5 %	5 %
	H.P. air source (climb and cruise condition)	3 %	3 %
	H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (maximum operating)

With fuel in the dorsal and/or ventral tank	280 knots
With dorsal and ventral tanks empty	320 knots
S.L. to 12,400 ft. decreasing linearly 1 kt. per 600 ft. to 292 kts. at 29,200 ft.	

V_{MO} (maximum operating) (Cont.)

With dorsal and ventral tanks empty and with Mod 258825: S.L. to 10,600 ft. decreasing linearly 1 kt. per 600 ft. to 288 kts. at 29,800 ft.	320 Knots

M_{MO} (maximum operating)

28,500 ft. and above	0.77 M

M_{MO} (maximum operating)(with Mod 252648)

29,200 ft. and above	0.78 M

V_A (maneuvering)

Sea level	192 knots
10,000 ft.	195 knots
20,000 ft.	198 knots
30,000 ft.	203 knots
35,000 ft.	207 knots
38,000 ft.	211 knots
40,000 ft.	214 knots
41,000 ft.	217 knots

V_{FE} (Flap speeds)

Deflection	
15^o	220 knots
25^o	175 knots
45^o	160 knots

V_{LO} (landing gear operation)

Retract	220 knots
Extend	220 knots

V_{LE} (landing gear extended) 220 knots

V_{MC} (minimum control speed)	APR not operating	APR operating
V_{MCA} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	101 knots	104 knots
V_{MCA} (with either rudder bias strut inoperative)	110 knots	113 knots
V_{MCG} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	92 knots	95 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC) 90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
24,800	25.00	(1.80)*	35.00	10.82
24,200	24.60	(1.44)*	-	-
24,200	21.80	1.09	36.80	12.45
22,000	-	-	36.80	12.45
20,950	-	-	36.27	11.97
20,850	18.60	3.98	33.80	9.74
20,650	-	-	33.67	9.62
20,400	-	-	30.27	6.56
19,000	-	-	29.40	5.77
16,300	18.00	4.52	-	-
14,700	-	-	31.93	8.05
13,100	18.00	4.52	31.50	7.67

*(Aft of Datum)
Straight line variations between weights

C.G. Range (with Modification 258332) (Gear and flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
25,500	27.40	(3.97)*	33.53	9.50
25,000	25.13	(1.92)*	-	-
24,200	24.60	(1.44)*	-	-
24,200	21.80	1.09	36.80	12.45
22,400	-	-	36.80	12.45
20,950	-	-	36.27	11.97
20,850	18.60	3.98	33.80	9.74
20,600	-	-	33.73	9.68
20,400	-	-	30.27	6.56
19,000	-	-	29.40	5.77
16,300	18.00	4.52	-	-
14,700	-	-	31.93	8.05
13,100	18.00	4.52	31.50	7.67

*(Aft of Datum)
Straight line variations between weights

Item (Extending)	Moment Change In. Lb.
Wing flaps 15°	+538
25°	+879
45°	+1593
Main landing gear	-1980
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		with Modification 258332	
Maximum ramp weight	25,000 Lbs.	25,500 Lbs.	(See NOTE 27)
Maximum brake release weight	24,800 Lbs.	25,500 Lbs.	(See NOTE 27)
Maximum landing weight	22,000 Lbs.	22,000 Lbs.	
Maximum zero fuel weight	16,050 Lbs.	16,300 Lbs.	(See NOTES 29 & 34)
Minimum zero fuel weight	13,100 Lbs.	13,100 Lbs.	

Minimum Crew For all flights, 2 pilots

Maximum Passengers 15

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward	180.25 to 223.11	109	310
Forward cabin			
(a) Side floor	245.85 to 303.85	50	
(b) Center floor	245.85 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	397.8 to 422.3	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	612.5	4,080	5.60
Tank 2	612.5	4,080	5.60
Engines and lines	1.5	10	107.60
Ventral tank	131.0	873	88.60
Dorsal tank	61.0	406	119.30
Total	1,418.5	9,449	18.26

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	93.69	1,059
No. 2	1.5	11.3	93.69	1,059
Total	3.0	22.6	93.90	2,118

Maximum Operating Altitude 41,000 feet

Serial Numbers Eligible 257001 through 257215 (See NOTE 61)

XI. Hawker Siddeley Model DH.125 Series 1A with Modifications 251867 and 252605 (Transport Aircraft), Approved January 20, 1982, (See NOTE 22).
The DH.125 Series 1A with modification 252605 aircraft differs respectively from the DH.125 Series 1A aircraft fitted with modification 251867 in the following major respects: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 521 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance.

Hawker Siddeley Model DH.125, Series 1A with Modification 252606 (Transport Aircraft), Approved January 20, 1982, (See NOTE 22)
The DH.125 Series 1A with modification 252606 aircraft differs respectively from the DH.125 Series 1A aircraft not fitted with modification 251867 in the following major respects: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 521 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance.

Engines

2 Garrett AiResearch TFE 731-3 turbofan engines, or
2 Garrett AiResearch TFE 731-3R turbofan engines (See NOTE 20)

Fuel

Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, 3-GP-23h, ASTM D.1655-74 Jet A or Jet A1
Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, ASTM D.1655 Jet B, MIL-T-5624 JP4 Grade, 3 GP-22 (See NOTE 28)

Engine Limits

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	3,700	3,880
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,700	3,700
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1) (21,000 rpm)	101.5 % (21,000 rpm)	101.5 %
H.P. Shaft (N2)	100 % (29,692 rpm)	101 % (29,989 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	907°C	929°C
Take-off (10 minutes maximum)	917°C	939°C
Take-off (instantaneous)	927°C	949°C
Maximum continuous	885°C	885°C
Engine starting and relighting (unrestricted)	907°C	907°C
Engine starting and relighting (10 seconds)	927°C	927°C
Engine starting and relighting (5 seconds)	above 927°C	above 927°C
Maximum permissible oil temperature:		
Sea level to 30,000 ft.	127°C	127°C
Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C

	Minimum permissible oil temperature:		
	Engine starting	-40°C	-40°C
	Before take-off	+30°C	+30°C
	Maximum permissible air bleed extraction:		
	L.P. air source	5 %	5 %
	H.P. air source (climb and cruise condition)	3 %	3 %
	H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (maximum operating) S.L. to 27,200 ft. decreasing linearly 1 kt. per 300 ft. to 273 kts. at 30,800 ft.	285 knots
M_{MO} (maximum operating)	0.755 M
V_A (Maneuvering)	
Sea level	185 knots
10,000 ft.	185 knots
20,000 ft.	185 knots
30,000 ft.	195 knots
V_A (Maneuvering)	
40,000 ft.	210 knots
Straight line variation between points shown.	
V_{FE} (Flap speeds)	
Deflection	
15°	210 knots
25°	160 knots
45°	145 knots
V_{LO} (landing gear operation)	
Retract	210 knots
Extend	210 knots
V_{LE} (landing gear extended)	210 knots

V_{MC} (minimum control speed)	APR not operating	APR operating
V_{MCA} (with flaps 0° or 15° at sea level for temperatures below 22°C	100 knots	104 knots
V_{MCA} (with either rudder bias strut inoperative)	110 knots	113 knots
V_{MCG} (with flaps 0° or 15° at sea level for temperatures below 22°C	91 knots	95 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

Series 1A with Modifications 251867 and 252605

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
21,700	24.60	(1.44)*	33.60	9.56
20,500	-	-	34.33	10.22
19,000	-	-	34.07	9.98
18,500	21.67	1.20	-	-
16,800	-	-	33.07	9.08
13,700	22.00	0.91	-	-
12,350	-	-	37.53	13.11
11,600	22.27	0.66	-	-
11,400	23.27	(0.24)*	37.53	13.11

*(Aft of Datum)

Straight line variations between weights

C.G. Range (Gear and Flaps Retracted)

Series 1A with Modification 252606

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
21,200	24.67	(1.50)*	33.80	9.74
20,500	-	-	34.33	10.22
19,000	-	-	34.07	9.98
18,000	21.67	1.20	-	-
16,800	-	-	33.07	9.08
13,700	22.00	0.91	-	-
12,350	-	-	37.53	13.11
11,600	22.20	0.73	-	-
11,400	23.27	(0.24)*	37.53	13.11

*(Aft of Datum)

Straight line variations between weights

Item (Extending)	Moment Change In. Lb.
Wing flaps 15^{o}	+538
25^{o}	+879
45^{o}	+1593
Main landing gear	-1800
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

	with Modifications 251867 & 252605	with Modification 252606
Maximum ramp weight	21,900 Lbs.	21,400 Lbs.
Maximum brake release weight	21,700 Lbs.	21,200 Lbs.
Maximum landing weight	19,550 Lbs.	19,550 Lbs.
Maximum zero fuel weight	13,700 Lbs.	13,200 Lbs.
Minimum zero fuel weight	11,400 Lbs.	11,400 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205.00 to 260.00	60	210
8 seater	205.00 to 260.00	60	160
Forward cabin			
(a) Side floor	260.00 to 303.85	50	
(b) Center floor	260.00 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	396.00 to 425.00	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Total	1,231.5	8,210	5.79

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	93.69	1,059
No. 2	1.5	11.3	93.69	1,059
Total	3.0	22.6		2,118

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — Same as listed previously for Hawker Siddeley Model DH.125 Series 1A and HS.125 Series 1B

XII. Hawker Siddeley Model DH.125 Series 3A with Modification 252603 (Transport Aircraft) Approved January 20, 1982 (See NOTE 26)
Hawker Siddeley Model HS.125 Series F3B (Transport Aircraft) Approved May 28, 1999. (See NOTES 26 and 52)
(The DH.125 Series 3A aircraft with modification 252603 and the HS.125 Series F3B differs respectively from the DH. 125 Series 3A and the HS.125 Series 3B aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance.

Engines — 2 Garrett AiResearch TFE 731-3 turbofan engines, or
2 Garrett AiResearch TFE 731-3R turbofan engines (See NOTE 20)

Fuel — Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, 3-GP-23h, ASTM D.1655-74 Jet A or Jet A1
Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, ASTM D.1655 Jet B, MIL-T-5624 JP4 Grade, 3 GP-22 (See NOTE 28)

Engine Limits

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	3,700	3,880
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,700	3,700
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	101.5 % (21,000 rpm)	101.5 % (21,000 rpm)
H.P. Shaft (N2)	100 % (29,692 rpm)	101 % (29,989 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	907°C	929°C
Take-off (10 minutes maximum)	917°C	939°C
Take-off (instantaneous)	927°C	949°C
Maximum continuous	885°C	885°C

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Engine starting and relighting (unrestricted)	907°C	907°C
Engine starting and relighting (10 seconds)	927°C	927°C
Engine starting and relighting (5 seconds)	above 927°C	above 927°C
Maximum permissible oil temperature:		
Sea level to 30,000 ft.	127°C	127°C
Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
Minimum permissible oil temperature:		
Engine starting	-40°C	-40°C
Before take-off	+30°C	+30°C
Maximum permissible air bleed extraction:		
L.P. air source	5 %	5 %
H.P. air source (climb and cruise condition)	3 %	3 %
H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (maximum operating) 285 knots
SL to 27,200 ft. decreasing linearly 1 kt. per 300 ft. to 273 kts. at 30,800 ft.

M_{MO} (maximum operating) 0.755 M

V_A (Maneuvering)

Sea level	185 knots
10,000 ft.	185 knots
20,000 ft.	185 knots
30,000 ft.	195 knots
40,000 ft.	210 knots

Straight line variation between points shown.

V_{FE} (Flap speeds)

Deflection	
15^o	210 knots
25^o	160 knots
45^o	145 knots

V_{LO} (landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (landing gear extended) 210 knots

Airspeed Limits (IAS) (Cont.)

V_{MC} (minimum control speed)	APR not operating	APR operating
V_{MCA} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	100 knots	104 knots
V_{MCA} (with either rudder bias strut inoperative)	110 knots	113 knots
V_{MCG} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	91 knots	95 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
21,700	24.60	(1.44)*	33.60	9.56
20,500	-	-	34.33	10.22
19,000	-	-	34.07	9.98
18,500	21.67	1.20	-	-
16,800	-	-	33.07	9.08
13,700	22.00	0.91	-	-
12,300	-	-	37.53	13.11
11,600	22.27	0.66	-	-
11,400	23.27	(0.24)*	37.53	13.11

*(Aft of Datum)
Straight line variations between weights

C.G. Range (Cont.)

Item (Extending)		Moment Change In. Lb.
Wing flaps	15°	+538
	25°	+879
	45°	+1593
Main landing gear		-1800
Nose landing gear		+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum ramp weight	21,900 Lbs.
Maximum brake release weight	21,700 Lbs.
Maximum landing weight	20,000 Lbs.
Maximum zero fuel weight	13,700 Lbs.
Minimum zero fuel weight	11,400 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205.00 to 260.00	60	210
8 seater	205.00 to 260.00	60	160
Forward cabin			
(a) Side floor	260.00 to 303.85	50	
(b) Center floor	260.00 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	395.00 to 425.00	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Total	1,231.5	8,210	5.79

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	93.69	1,059
No. 2	1.5	11.3	93.69	1,059
Total	3.0	22.6		2,118

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible

Same as listed previously for Hawker Siddeley Model DH.125 Series 3A and HS.125 Series 3B (See NOTE 61)

XIII. Hawker Siddeley Model DH.125, Series 3A/RA with Modification 252600 (Transport Aircraft) Approved February 15, 1968 (See NOTE 25)
Hawker Siddeley Model HS.125 Series F3B/RA (Transport Aircraft) Approved May 28, 1999. (See NOTES 25 and 52).
(The DH.125 Series 3A/RA aircraft with modification 252600 and the HS.125 Series F3B/RA differs respectively from the DH.125 Series 3A/RA and the HS.125 Series 3B/RA aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance.

Engines

2 Garrett AiResearch TFE 731-3 turbofan engines, or
2 Garrett AiResearch TFE 731-3R turbofan engines (See NOTE 20)

Fuel

Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, 3-GP-23h, ASTM.D.1655-74 Jet A or Jet A-1
Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, ASTM.D.1655 Jet B, MIL-T-5624 JP4 Grade, 3 GP-22 (See NOTE 28)

Engine Limits

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	3,700	3,880
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,700 3,700	
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	101.5 % (21,000 rpm)	101.5 % (21,000 rpm)
H.P. Shaft (N2)	100 % (29,692 rpm)	100 % (29,989 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	907°C	929°C
Take-off (10 minutes maximum)	917°C	939°C
Take-off (instantaneous)	927°C	949°C
Maximum continuous	885°C	885°C
Engine starting and relighting (unrestricted)	907°C	907°C
Engine starting and relighting (10 seconds)	927°C	927°C
Engine starting and relighting (5 seconds)	above 927°C	above 927°C
Maximum permissible oil temperature:		
Sea level to 30,000 ft.	127°C	127°C
Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
Minimum permissible oil temperature:		
Engine starting	-40°C	-40°C
Before take-off	+30°C	+30°C

Maximum permissible air bleed extraction:

L.P. air source	5 %	5 %
H.P. air source (climb and cruise condition)	3 %	3 %
H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (Maximum operating)

with fuel in long range tanks 257 knots

from sea level to 27,500 feet with long range tank empty 282 knots

decreasing linearly 1 knot per 320 feet to 270 knots at 31,350 feet.

M_{MO} (maximum operating) 0.755 M

V_A (Maneuvering) 193 knots

V_{FE} (Flap speeds)

Deflection	
15^o	210 knots
25^o	160 knots
45^o	145 knots

V_{LO} (landing gear operation)

Retract	210 knots
Extend	210 knots

V_{LE} (landing gear extended) 210 knots

V_{MC} (minimum control speed)	APR not operating	APR operating
V_{MCA} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	100 knots	104 knots
V_{MCA} (with either rudder bias strut inoperative)	110 knots	113 knots
V_{MCG} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	91 knots	95 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,600	30.73	(6.97)*	36.00	11.73
23,300	-	-	37.00	12.63
22,900	30.40	(6.67)*	-	-
22,900	27.40	(3.97)*	-	-
21,900	-	-	38.00	13.53
19,600	-	-	38.00	13.53
19,300	-	-	34.07	9.98
18,500	23.87	(0.78)*	-	-
17,400	-	-	33.27	9.26
16,100	23.23	(0.20)*	-	-
14,700	26.20	(2.88)*	-	-
14,200	-	-	36.13	11.84
13,750	25.00	(1.80)*	-	-
13,000	-	-	37.53	13.11
11,600	25.00	(1.80)*	-	-
11,400	26.00	(2.70)*	37.53	13.11

*(Aft of Datum)

Straight line variations between weights

Item (Extending)	Moment Change In. Lb.
Wing flaps 15°	+538
25°	+879
45°	+1593
Main landing gear	-1800
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum ramp weight	23,800 Lbs.
Maximum brake release weight	23,600 Lbs.
Maximum landing weight	20,000 Lbs.
Maximum zero fuel weight	14,700 Lbs.
Minimum zero fuel weight	11,400 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205.00 to 260.00	60	210
8 seater	205.00 to 260.00	60	160
Forward cabin			
(a) Side floor	260.00 to 303.85	50	
(b) Center floor	260.00 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	395.00 to 425.00	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Long Range Tank	134.5	896	88.70
Total	1,366.0	9,106	13.95

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	93.69	1,059
No. 2	1.5	11.3	93.69	1,059
Total	3.0	22.6		2,118

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible

Same as listed previously for Hawker Siddeley Model DH.125 Series 3A and HS.125 Series 3B (See NOTE 61)

XIV. Hawker Siddeley Model DH.125, Series 400A with Modification 252550 Approved November 15, 1968 (Transport Aircraft) (See NOTE 21)
Beechcraft Hawker Model BH.125, Series 400A with Modification 252550 Approved July 14, 1970 (Transport Aircraft) (See NOTE 21)
(The DH.125 Series 400A aircraft with modification 252550 and the BH.125 Series 400A aircraft with modification 252550 differs respectively from the DH.125 Series 400A and the BH.125 Series 400A aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance, and (iv) Increase in certificated taxi and take-off weights.

Hawker Siddeley Model HS.125 Series F400B (Transport Aircraft) Approved May 28, 1999. (See NOTES 21 and 53).
(The HS.125 Series F400B aircraft with modification 252551 differs from the HS.125 Series 400B aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance, and (iv) Increase in certificated taxi and take-off weights.

Hawker Siddeley Model HS.125 Series F403B (Transport Aircraft) Approved May 28, 1999. (See NOTES 21 and 53).
(The HS.125 Series F403B aircraft with modification 252551 differs from the HS.125 Series 403B aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance, and (iv) Increase in certificated taxi and take-off weights.

Engines
2 Garrett AiResearch TFE 731-3 turbofan engines, or
2 Garrett AiResearch TFE 731-3R turbofan engines (See NOTE 20)

Fuel
Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, 3-GP-23h, ASTM.D.1655-74 Jet A or Jet A-1
Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, ASTM.D.1655 Jet B, MIL-T-5624 JP4 Grade, 3 GP-22 (See NOTE 28)

Engine Limits

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	3,700	3,880
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,700	3,700
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	101.5 % (21,000 rpm)	101.5 % (21,000 rpm)
H.P. Shaft (N2)	100 % (29,692 rpm)	100 % (29,989 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	907°C	929°C
Take-off (10 minutes maximum)	917°C	939°C
Maximum permissible interstage turbine temperature (ITT): (Cont.)		
Take-off (instantaneous)	927°C	949°C
Maximum continuous	885°C	885°C
Engine starting and relighting unrestricted)	907°C	907°C
Engine starting and relighting (10 seconds)	927°C	927°C
Engine starting and relighting (5 seconds)	above 927°C	above 927°C

Engine Limits (Cont.)

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Maximum permissible oil temperature:		
Sea level to 30,000 ft.	127°C	127°C
Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
Minimum permissible oil temperature:		
Engine starting	-40°C	-40°C
Before take-off	+30°C	+30°C
Maximum permissible air bleed extraction:		
L.P. air source	5 %	5 %
H.P. air source (climb and cruise condition)	3 %	3 %
H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (Maximum operating)	
with fuel in long range tanks	257 knots
from sea level to 27,500 feet with long range tank empty	282 knots
decreasing linearly 1 knot per 320 feet to	270 knots at 31,350 feet.
V_{MO} (Maximum operating) (with modification 259273)	
with fuel in long range tanks	253 knots
from sea level to 27,500 feet with long range tank empty	276 knots
decreasing linearly 1 knot per 340 feet to	260 knots at 32,940 feet.
M_{MO} (maximum operating)	0.755 M
V_A (Maneuvering)	193 knots
V_{FE} (Flap speeds)	Deflection
15°	210 knots
25°	160 knots
45°	145 knots
V_{LO} (landing gear operation)	
Retract	210 knots
Extend	210 knots
V_{LE} (landing gear extended)	210 knots

V_{MC} (minimum control speed)	APR not operating	APR operating
V_{MCA} (with flaps 0° or 15° at sea level for temperatures below 22°C	100 knots	104 knots
V_{MCA} (with either rudder bias strut inoperative)	110 knots	113 knots
V_{MCG} (with flaps 0° or 15° at sea level for temperatures below 22°C	91 knots	95 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
23,600	30.73	(6.97)*	36.00	11.73
23,300	-	-	37.00	12.63
22,900	30.40	(6.67)*	-	-
22,900	27.40	(3.97)*	-	-
21,900	-	-	38.00	13.53
19,600	-	-	38.00	13.53
19,300	-	-	34.07	9.98
18,500	23.87	(0.78)*	-	-
17,400	-	-	33.27	9.26
16,100	23.23	(0.20)*	-	-
14,700	26.20	(2.88)*	-	-
14,200	-	-	36.13	11.84
13,750	25.00	(1.80)*	-	-
13,000	-	-	37.53	13.11
11,600	25.00	(1.80)*	-	-
11,400	26.00	(2.70)*	37.53	13.11

*(Aft of Datum)

Straight line variations between weights

Item (Extending)	Moment Change In. Lb.
Wing flaps 15^{o}	+538
25^{o}	+879
45^{o}	+1593
Main landing gear	-1800
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		with Modification 259273
Maximum ramp weight	23,800 Lbs.	23,800 Lbs.
Maximum brake release weight	23,600 Lbs.	23,600 Lbs.
Maximum landing weight	20,000 Lbs.	20,000 Lbs.
Maximum zero fuel weight	14,700 Lbs.	15,200 Lbs.
Minimum zero fuel weight	11,400 Lbs.	11,400 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

8

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward			
6 seater	205.00 to 260.00	60	210
8 seater	205.00 to 260.00	60	160
Forward cabin			
(a) Side floor	260.00 to 303.85	50	
(b) Center floor	260.00 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	395.00 to 425.00	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	615.0	4,100	5.70
Tank 2	615.0	4,100	5.70
Engines and lines	1.5	10	81.00
Long Range Tank	134.5	896	88.70
Total	1,366.0	9,106	13.95

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	93.69	1,059
No. 2	1.5	11.3	93.69	1,059
Total	3.0	22.6		2,118

Maximum Operating Altitude 40,000 feet (See NOTE 9)

Serial Numbers Eligible — Same as listed previously for Hawker Siddeley Models DH.125 Series 400A and HS.125 Series 400B and Beechcraft Hawker Model BH.125 Series 400A. (See NOTE 61)

XV. Beechcraft Hawker Model BH.125 Series 600A with Modification 252468 (Transport Aircraft) Approved October 15, 1981 (See NOTE 19)

Hawker Siddeley Model HS.125 Series 600A with Modification 252468 (Transport Aircraft) Approved October 15, 1981. (See NOTE 19)

The BH.125 Series 600A aircraft with modification 252468 and the HS.125 Series 600A aircraft with modification 252468 differs respectively from the BH.125 Series 600A and the HS.125 Series 600A aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance, and (iv) Reduction in take-off weights and increase in maximum zero fuel weight.

The BH.125 Series 600A with modification 252468, is, from aircraft Serial No. 256055, identified as the model HS.125 Series 600A with modification 252468.

Hawker Siddeley Model HS.125 Series F600B (Transport Aircraft) Approved May 28, 1999. (See NOTES 19 and 54).

The HS.125 Series F600B aircraft differs from the HS.125 Series 600B aircraft in the following major features: (i) Garrett AiResearch TFE 731-3 engines replace the Rolls Royce Viper 522 turbine engines, (ii) Modifications to associated aircraft systems consequential to the engine change, (iii) Minor changes to improve aerodynamic efficiency and aircraft appearance, and (iv) Reduction in take-off weights and increase in maximum zero fuel weight.

Engines — 2 Garrett AiResearch TFE 731-3 turbofan engines, or
2 Garrett AiResearch TFE 731-3R turbofan engines (See NOTE 20)

Fuel

Aviation Kerosene to specification Defence Standard 91-91, NATO Code F-35, Defence Standard 91-87, NATO Code F-34, 3-GP-23h Type 1, ASTM D.1655 Jet A or Jet A1. Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, 3-GP-22 Type 2, ASTM D.1655 Jet B, Mil-T-5624 JP4. (See NOTE 28).

Engine Limits

	TFE 731-3 and TFE 731-3R with APR not operating	TFE 731-3R with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	3,700	3,880
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	3,700	3,700
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	101.5 % (21,000 rpm)	101.5 % (21,000 rpm)
H.P. Shaft (N2)	100 % (29,692 rpm)	100 % (29,989 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	907°C	929°C
Take-off (10 minutes maximum)	917°C	939°C
Take-off (instantaneous)	927°C	949°C
Maximum continuous	885°C	885°C
Engine starting and relighting (unrestricted)	907°C	907°C
Engine starting and relighting (10 seconds)	927°C	927°C
Engine starting and relighting (5 seconds)	above 927°C	above 927°C
Maximum permissible oil temperature: Sea level to 30,000 ft.	127°C	127°C
Maximum permissible oil temperature: Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
Minimum permissible oil temperature:		
Engine starting	-40°C	-40°C
Before take-off	+30°C	+30°C
Maximum permissible air bleed extraction:		
L.P. air source	5 %	5 %
H.P. air source (climb and cruise condition)	3 %	3 %
H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (maximum operating)

With fuel in the dorsal and/or ventral tank	280 knots
With dorsal and ventral tanks empty	320 knots
S.L. to 12,400 ft. decreasing linearly 1 kt. per 600 ft. to 292 kts. at 29,200 ft.	

M_{MO} (maximum operating)

28,500 ft. and above	0.78 M

V_A (maneuvering)

Sea level	192 knots
10,000 ft.	195 knots
20,000 ft.	198 knots
30,000 ft.	203 knots
35,000 ft.	207 knots
38,000 ft.	211 knots
40,000 ft.	214 knots
41,000 ft.	217 knots

V_{FE} (Flap speeds)

Deflection	
15^o	220 knots
25^o	175 knots
45^o	160 knots

V_{LO} (landing gear operation)

Retract	220 knots
Extend	220 knots

V_{LE} (landing gear extended) 220 knots

V_{MC} (minimum control speed)	APR not operating	APR operating
V_{MCA} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	101 knots	104 knots
V_{MCA} (with either rudder bias strut inoperative)	110 knots	113 knots
V_{MCG} (with flaps 0^o or 15^o at sea level for temperatures below 22^oC	92 knots	95 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

90.24 in. The leading edge of the SMC is 20.76. in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
24,800	25.00	(1.80)*	35.00	10.82
24,200	24.60	(1.44)*	-	-
24,200	21.80	1.09	36.47	12.15
22,500	-	-	36.47	12.15
20,950	-	-	36.27	11.97
20,850	18.60	3.98	33.80	9.74
20,650	-	-	33.67	9.62
20,400	-	-	30.27	6.56
19,000	-	-	29.40	5.77
16,050	18.00	4.52	-	-
14,700	-	-	31.93	8.05
13,100	18.00	4.52	31.50	7.66

*(Aft of Datum)

Straight line variations between weights

C.G. Range (with Modification 252818) (Gear and flaps Retracted)

Wt. Lbs.	Fwd. of Datum % SMC	Fwd. of Datum In.	Aft of Datum % SMC	Aft of Datum In.
25,500	27.13	(3.72)*	33.53	9.50
25,100	25.00	(1.80)*	-	-
24,450	24.60	(1.44)*	-	-
24,450	21.80	1.07	-	-
24,200	-	-	36.47	12.15
22,400	-	-	36.47	12.15
21,100	18.60	3.98	-	-
20,950	-	-	36.27	11.97
20,875	-	-	33.80	9.74
20,600	-	-	33.73	9.68
20,400	-	-	30.33	6.61
19,000	-	-	29.40	5.77
16,300	18.00	4.52	-	-
14,700	-	-	31.87	8.00
13,100	18.00	4.52	31.50	7.66

*(Aft of Datum)
Straight line variations between weights

Item (Extending)	Moment Change In. Lb.
Wing flaps 15^{o}	+538
25^{o}	+879
45^{o}	+1593
Main landing gear	-1980
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		with Modification 252818	
Maximum ramp weight	25,000 Lbs.	25,500 Lbs.	(See NOTE 27)
Maximum brake release weight	24,800 Lbs.	25,500 Lbs.	(See NOTE 27)
Maximum landing weight	22,000 Lbs.	22,000 Lbs.	
Maximum zero fuel weight	16,050 Lbs.	16,050 Lbs.	
Minimum zero fuel weight	13,100 Lbs.	13,100 Lbs.	

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

15

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward	180.25 to 223.11	100	310
Forward cabin			
(a) Side floor	245.85 to 303.85	50	
(b) Center floor	245.86 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	397.80 to 422.30	60	130

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	612.5	4,080	5.60
Tank 2	612.5	4,080	5.60
Engines and lines	1.5	10	107.60
Ventral tank	131.0	873	88.60
Dorsal tank	61.0	406	119.30
Total	1,418.5	9,449	18.26

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	93.69	1,059
No. 2	1.5	11.3	93.69	1,059
Total	3.0	22.6	93.90	2,118

Maximum Operating Altitude. 41,000 feet

Serial Numbers Eligible. Same as listed previously for models BH/HS.125 Series 600A and HS.125 Series 600B, 600B/1, 600B/2 and 600B/3

XVI. British Aerospace Model BAe.125 Series 800A (Transport Aircraft) Approved July 12, 1984 (See NOTES 30, 40 and 64)

British Aerospace Model BAe.125 Series 800A (C-29A) (Transport Aircraft) Approved December 12, 1989 (See NOTES 30 and 64)

The C-29A variant was intended for Airborne Flight Inspection operations.

British Aerospace Model BAe.125 Series 800A (U-125)(Transport Aircraft) Approved April 24, 1992 (See NOTES 40 and 64)

The U-125 variant was intended for Airborne Flight Inspection operations.

British Aerospace Model BAe.125 Series 800B (Transport Aircraft) Approved May 28, 1999. (See NOTE 57)

Hawker 800 (name change) (Transport Aircraft)Approved January 28, 1994 (See NOTES 42, 57 and 64)

The BAe.125 Series 800A/Hawker 800 and the BAe.125 Series 800B differs respectively from the HS.125 Series 700A and the HS.125 Series 700B aircraft in the following major respects: (i) Garrett Turbine Engine Company TFE 731-5R engines replace the Garrett AiResearch TFE 731-3 engines, (ii) The wing span is increased by 4 ft. 6 ins, (iii) Curved windscreens replace the existing flat panels, (iv) The rear fuselage underfairing is reshaped and the ventral tank is increased in capacity. The ventral fin is deleted, (v) The fin leading edge is extended forward and the dorsal fuel tank deleted, (vi) The nose wheel doors are sequenced to close after the gear is down, (vii) A stall identification (stick pusher) system is fitted, (viii) An Electronic Flight Instrument System (E.F.I.S.) is fitted, (ix) Increase in certificated taxiing, take-off, landing and zero fuel weights, and (x) Increase of M_{MO} from 0.77 to 0.80. (See NOTE 36 and 42).

Hawker 800 (U-125A) (Transport Aircraft) Approved December 9, 1994 (See NOTES 49 and 64)

The U-125A variant was intended for use by the Japan Air Self Defense Force as a search and rescue aircraft.

Engines 2 Garrett Turbine Engine Company TFE 731-5R turbofan engines

Fuel Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, ASTM .D.1655 Jet A or Jet A-1, CAN/CGSB 3.23/, MIL-T-83133 JP8 Grade.

Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, ASTM D.1655 Jet B, MIL-T-5624 JP4 and JP5 Grades, CAN/CGSB 3.22/ Jet B, GOST 10227-86 T-2 (See NOTE 28)

Engine Limits		TFE 731-5R with APR not operating	TFE 731-5R with APR operating
	Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	4,304	4,500
	Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	4,304	4,304
	Maximum permissible engine rotor operating speed		
	L.P. Shaft (N1)	100 % (21,000 rpm)	100 % (21,000 rpm)
	H.P. Shaft (N2)	100 % (29,692 rpm)	100 % (29,989 rpm)
	Maximum permissible interstage turbine temperature (ITT):		
	Take-off (5 minutes maximum)	952°C	974°C
	Take-off (10 minutes maximum)	984°C	984°C
	Take-off (instantaneous)	994°C	994°C
	Maximum continuous	924°C	924°C
	Engine starting and relighting (unrestricted)	952°C	952°C
	Engine starting and relighting (10 seconds)	974°C	974°C
	Engine starting and relighting (5 seconds)	above 974°C	above 974°C
	Maximum permissible oil temperature:		
	Sea level to 30,000 ft.	127°C	127°C
	Above 30,000 ft.	140°C	140°C
	Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
	Minimum permissible oil temperature:		
	Engine starting	-40°C	-40°C
	Before take-off	+30°C	+30°C
	Maximum permissible air bleed extraction:		
	L.P. air source	5 %	5 %
	H.P. air source (climb and cruise condition)	3 %	3 %
	H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)		
	V_{MO} (maximum operating)	
	With fuel in the ventral tank	280 knots
	With ventral tank empty or with the pannier fitted to BAe. Mod 259292 (See NOTE 33) S.L. to 12,000 ft. decreasing linearly 1 kt. per 680 ft. to 310 kts. at 29,000 ft.	335 knots
	V_{MO} (maximum operating) (with Mod. 25B047A) S.L. to 12,000 ft. decreasing linearly 1 kt. per 680 ft. to 313 kts. at 27,300 ft	335 knots
	M_{MO} (maximum operating)	0.80 M
	M_{MO} (maximum operating) (with Mod. 25B047A)	0.78 M

V_A (maneuvering)	
Sea level	196 knots
10,000 ft.	202 knots
20,000 ft.	207 knots
30,000 ft.	217 knots
35,000 ft.	225 knots
38,000 ft.	231 knots
40,000 ft.	236 knots
41,000 ft.	238 knots
V_{FE} (Flap speeds)	
Deflection	
15^o	220 knots
25^o	175 knots
45^o	165 knots
V_{LO} (landing gear operation)	
Retract	220 knots
Extend	220 knots
V_{LE} (landing gear extended)	220 knots
V_{MC} (minimum control speed)	
V_{MCA} (with flaps 0^o or 15^o at sea level for temperatures below 23^oC	115 knots
V_{MCA} (with either rudder bias strut inoperative)	125 knots
V_{MCG} (with flaps 0^o or 15^o at sea level for temperatures below 23^oC	112 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

87.16 in. The leading edge of the SMC is 15.70 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
27,400	25.60	(6.61)*	31.80	12.02
26,600	-	-	35.00	14.81
26,000	23.70	(4.96)*	-	-
25,800	19.20	(1.03)*	-	-
25,000	17.60	0.36	-	-
24,000	16.40	1.40	35.00	14.81
23,000	15.60	2.10	-	-
22,600	-	-	33.20	13.24
22,400	-	-	28.00	8.70
22,000	15.20	2.45	-	-
21,400	-	-	26.20	7.14
21,000	15.00	2.63	-	-
20,000	-	-	26.60	7.48
19,000	-	-	27.30	8.09
18,000	-	-	28.20	8.88
17,520	15.70	2.02	24.60	5.74
17,200	-	-	29.20	9.75
16,600	-	-	29.60	10.10
15,750	15.00	2.63	-	-
15,400	15.70	2.02	-	-
14,120	19.00	(0.86)*	28.60	9.23

*(Aft of Datum)

C.G. Range (with Modifications 259550 & 253169A) (Gear and Flaps Retracted)

Wt. Lbs.	Fwd. of Datum % SMC	Fwd. of Datum In.	Aft of Datum % SMC	Aft of Datum In.
28,000	25.60	(6.61)*	32.00	12.19
27,150	-	-	35.00	14.81
26,600	23.50	(4.78)*	-	-
26,500	19.50	(1.30)*	-	-
26,000	18.00	0.00	-	-
25,000	16.80	1.06	-	-
24,000	16.10	1.67	35.00	14.81
23,000	15.55	2.15	-	-
22,600	-	-	33.10	13.15
22,400	-	-	28.40	9.05
22,000	15.20	2.45	-	-
21,400	15.00	2.63	26.30	7.22
20,300	-	-	26.60	7.48
18,000	15.70	2.02	24.60	5.74
17,600	-	-	29.20	9.75
17,000	-	-	29.60	10.10
15,600	15.00	2.63	-	-
15,400	15.70	2.02	-	-
14,120	19.00	(0.86)*	28.60	9.23

*(Aft of Datum)

Straight line variations between weights.

C.G. Range (with Mod. 25B047A) (Gear and Flaps Retracted)

Wt. Lbs.	Fwd. of Datum % SMC	Fwd. of Datum In.	Aft of Datum % SMC	Aft of Datum In.
26,866	19.00	(0.86)*	23.30	.61
25,000	16.20	1.58	24.50	5.65
22,000	15.00	2.63	21.30	2.86
19,500	15.00	2.63	-	-
19,200	-	-	21.30	2.86
18,450	15.80	1.93	21.50	3.04
17,350	15.80	1.93	-	-
17,000	-	-	23.00	4.35
16,550	-	-	23.00	4.35

*(Aft of Datum)

Straight line variations between weights.

Item (Extending)	Moment Change In. Lb.
Wing flaps 15^{o}	+538
25^{o}	+879
45^{o}	+1593
Main landing gear	-1980
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		with Modifications 259550 & 253169A	
Maximum ramp weight	27,520 Lbs.	28,100 Lbs.	(See NOTE 31)
Maximum brake release weight	27,400 Lbs.	28,000 Lbs.	(See NOTE 31)
Maximum landing weight	23,350 Lbs.	23,350 Lbs.	
Maximum zero fuel weight	17,520 Lbs.	18,000 Lbs.	(See NOTES 32 & 37)
Minimum zero fuel weight	14,120 Lbs.	14,120 Lbs.	

	with Modification 25B047A
Maximum ramp weight	26,866 Lbs.
Maximum brake release weight	26,866 Lbs.
Maximum landing weight	23,350 Lbs.
Maximum zero fuel weight	18,450 Lbs.
Minimum zero fuel weight	16,550 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

15

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward	180.25 to 223.11	100	310
Forward cabin			
(a) Side floor	245.85 to 303.85	50	
(b) Center floor	245.86 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	397.80 to 422.30	50	60

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	631.75	4208	8.20
Tank 2	631.75	4208	8.20
Ventral tank	231.80	1,544	100.40
Total	1,495.30	9,960	22.49

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.5	11.3	90.84	1,026
No. 2	1.5	11.3	90.84	1,026
Total	3.0	22.6	90.84	2,052

Maximum Operating Altitude.

41,000 feet

Serial Numbers Eligible.

BAe.125 Series 800A and 800B - 258001 through 258128, 258130, 258132, 258133, 258135 through 258150, 258152, 258153, 258155, 258157, 258160 through 258214, 258216 through 258226, 258228 through 258241, 258243, 258244, 258246, 258248, 258249, 258251 through 258254, (See NOTE 61);

C-29A - 258129, 258131, 258134, 258154, 258156 and 258158

U-125 - 258215, 258227 and 258242

Hawker 800 - 258255 through 258265, 258267, 258269 through 258276

U-125A - 258245, 258247, 258250, 258268, 258288, 258305, 258306, 258325, 258333, 258341, 258348, 258360, 258370, 258381, 258407, 258427, 258445, 258469, 258493, 258513 and 258533

XVII. British Aerospace Model BAe.125 Series 1000A (Transport Aircraft) Approved October 31, 1991. (See NOTES 39, 45 and 64)
British Aerospace Model BAe.125 Series 1000B (Transport Aircraft) Approved May 28, 1999. (See NOTES 45 and 58)
Hawker 1000 (name change) Approved January 28, 1994.(See NOTES 41, 45 and 64)
The BAe.125 Series 1000A, 1000B and Hawker 1000 differs respectively from the BAe.125 Series 800A, 800B and Hawker 800 aircraft in the following major respects, (i) Pratt and Whitney Canada (P&WC) PW305 engines with Full Authority Digital Engine Control replace the Garrett Turbine Engine Company TFE 731-5R Engines, (ii) Fuselage is lengthened by 33 inches and the addition of a window on each side, (iii) A forward ventral tank is introduced and the aft ventral tank increased in capacity, (iv) An external rear baggage compartment loading door is introduced (See NOTE 45), (v) Split pitch and roll control systems are introduced, (vi) A secondary pressure bulkhead is introduced between the toilet and the rear baggage, (vii) An increase in the Maximum Operating Altitude to 43,000 feet is introduced, and (vii) An increase in certificated taxiing, take-off, landing and zero fuel weights are introduced.

Engines

2 Pratt & Whitney Canada (P&WC) PW305 turbofan engines, or
2 Pratt & Whitney Canada (P&WC) PW305B turbofan engines (Post Mod. 253650A) (See NOTE 43)

Fuel

Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, ASTM .D.1655 (Jet A or Jet A-1), MIL-T-5624 - JP4 and JP5 Grades, MIL-T-83133 JP8 Grade, CAN/CGSB 3.23 (Jet A or A-1), GOST 10227-86 (TS-1 premium and RT.)

Engine Limits

	PW305 engine	PW305B engine
Normal take-off static thrust (to 15^{O}C OAT) Lbs.	5,225	5,204
Maximum take-off static thrust (to 22^{O}C OAT) Lbs.	5,225	-
Maximum take-off static thrust (to 23.5^{O}C OAT) Lbs.	-	5,266
Maximum continuous thrust (to 19.4^{O}C OAT) Lbs.	4,750	-
Maximum continuous thrust (to 27.5^{O}C OAT) Lbs.	-	4,483
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	102 % (10,820rpm)	102 % (10,820 rpm)
H.P. Shaft (N2)	102 % (27,469rpm)	102 % (47,469 rpm)
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	785^{O}C	785^{O}C
Take-off (20 seconds maximum)	795^{O}C	795^{O}C
Maximum continuous	785^{O}C	785^{O}C
Engine starting and relighting (unrestricted)	680^{O}C	680^{O}C
Maximum permissible oil temperature:		
Maximum oil temperature.	135^{O}C	135^{O}C
Transient limit (20^{O}C)	143^{O}C	143^{O}C

	Minimum permissible oil temperature:		
	Engine starting	-40°C	-40°C
	Before take-off	+10°C	+10°C
	Maximum permissible air bleed extraction:		
	L.P. air source	5 %	5 %
	H.P. air source (climb and cruise condition)	7 %	7 %
	Combined total air source	10 %	10 %
Airspeed Limits (IAS)	V_{MO} (Maximum operating) with ventral tanks empty, up to 12,900 feet reducing by 1 kt. per 750 ft to 308 knots at 29,400 ft.	330 knots	
	with ventral tanks not empty up to an altitude of 33,730 feet.	280 knots	
	M_{MO} (maximum operating) 0.80 M		
	V_A (Maneuvering)	200 knots	
	V_{FE} (Flap speeds)		
	Deflection		
	15°	220 knots	
	25°	180 knots	
	45°	170 knots	
	V_{LO} (Landing gear operation)		
	Retract	220 knots	
	Extend	220 knots	
	V_{LE} (Landing gear extended)	220 knots	
		with Modification 253650A	
	V_{MC} (Minimum control speed)		
	V_{MCG} (with flaps 0° or 15° at sea level for temperatures below 20°C	119 knots	119.5 knots
	V_{MCA} (with flaps at 0° at sea level for temperatures below 20°C)	124 knots	116.5 knots
	V_{MCA} (with flaps at 15° at sea level for temperatures below 20°C)	119 knots	111.5 knots

Datum

The center of gravity datum (station 353.04 inches) is 12.25 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Cord (SMC)

87.16 in. The leading edge of the SMC is 15.70 in., forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
31,000	22.00	(3.48)*	31.00	11.32
30,300	-	-	33.40	13.41
29,000**	12.50	4.81	19.30	1.12
27,000	-	-	34.00	13.93
26,500**	16.00	1.75	-	-
24,000**	12.50	4.81	15.40	(2.24)*X
22,500	-	-	31.40	11.67
22,000	15.00	2.63	-	-
19,800***	16.18	1.60	28.42	7.66
19,000	15.00	2.63	-	-
19,000***	-	-	31.00	11.32
16,300	16.00	1.75	-	-
16,300***	16.00	1.75	-	-
15,800***	20.80	(2.43)*	31.00	11.32
15,800	20.80	(2.43)*	33.30	13.32

*(Dimension Aft of Datum)
**(Boundary Area for Fuel Transfer In-Flight only)
***(Boundary Area for Zero Fuel Weight)
*X(Dimension Fwd of Datum)
Straight line variations between weights.

C.G. Range (Gear and Flaps Retracted) (with Modification 253379A)

	Fwd. of Datum		Aft of Datum	
Wt. Lbs.	% SMC	In.	% SMC	In.
31,000	22.00	(3.48)*	31.00	11.32
30,300	-	-	33.40	13.41
29,000**	12.50	4.81	19.30	1.12
27,000	-	-	34.00	13.93
26,500	16.00	1.75	-	-
26,500**	16.00	1.75	-	-
24,000**	12.50	4.81	15.40	(2.24)*X
22,500	-	-	31.40	11.67
22,000	15.00	2.63	-	-
20,300***	16.20	1.58	26.80	7.66
19,000	15.00	2.63	-	-
19,000***	-	-	31.00	11.32
16,300	16.00	1.75	-	-
16,300***	16.00	1.75	-	-
15,800***	20.80	(2.43)*	31.00	11.32
15,800	20.80	(2.43)*	33.30	13.32

*(Dimension Aft of Datum)

C.G. Range (Gear and Flaps Retracted) (with Modification 253379A)(Cont.)

**(Boundary Area for Fuel Transfer In-Flight only)
***(Boundary Area for Zero Fuel Weight)
*X(Dimension Fwd of Datum)
Straight line variations between weights.

Item (Extending)	Moment Change In. Lb.
Wing flaps 15°	+538
25°	+879
45°	+1593
Main landing gear	-1980
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

		with Modification 253379A
Maximum ramp weight	31,100 Lbs.	31,100 Lbs.
Maximum brake release weight	31,000 Lbs.	31,000 Lbs.
Maximum landing weight	25,000 Lbs.	25,000 Lbs.
Maximum zero fuel weight	19,800 Lbs.	20,300 Lbs. (See NOTES 38 & 44)
Minimum zero fuel weight	15,800 Lbs.	15,800 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

15

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward Stowage Wardrobe	107.35 to 192.25	100	80
Forward Cabin			
(a) Side Floor	214.75 to 303.85	50	-
(b) Center floor	214.75 to 303.85	60	-
Aft Cabin			
(a) Side floor	303.85 to 381.75	50	-
(b) Center floor	303.85 to 381.75	60	-
Wardrobe	370.81 to 381.56	100	20
Aft Luggage Compartment	410.30 to 445.40	100	500

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	637.0	4243	8.24
Tank 2	637.0	4243	8.24
Fwd Ventral tank	164.0	1092	-87.78
Aft Ventral tank*	270.0	1798	109.07
Total	1,708.0	11,376	15.00

Usable Fuel (pressure refueled)

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	634.6	4227	8.24
Tank 2	634.6	4227	8.24
Fwd Ventral tank	160.4	1068	-87.78
Aft Ventral tank*	266.4	1774	109.07
Total	1,696.0	11,296	15.00

* If external toilet servicing facility is fitted See NOTE 62.

Oil Capacity

The oil tank has a capacity of 2.0 U.S. gallons of oil, of which 1.25 U.S. gallons may be consumed without adversely affecting the operation of the engine. The engine oil tank in the BAe.125 Series 1000A, 1000B and Hawker 1000 is an integral part of the engine.

Maximum Operating Altitude.

43,000 feet

Serial Numbers Eligible.

BAe.125 Series 1000A and 1000B: 258151, 258159, 259004 through 259042. (See NOTE 61)
Hawker 1000 - 259003, 259043 through 259052

XVIII. Hawker 800XP (Transport Aircraft) Approved July 28, 1995 (See NOTES 47, 59, 64 and 65).

The Hawker 800XP differs respectively from the BAe.125 Series 800A aircraft in the following major respects: (i) Allied Signal Engines TFE 731-5BR turbofan engines replace the Garrett Turbine Engine Company TFE 731-5R turbofan engines, (ii) Dee Howard TR5000BR thrust reversers fitted as standard, (iii) Increase in certificated ramp, take-off and maximum zero fuel weights, (iv) Vortilons replace wing fences and Hawker 1000 aileron servo tab gearing is introduced, (v) Rudder Bias moment arm is reduced to 2.72", (vi) Mach Trim System is fitted, (vii) 3 Wheel ECS is fitted as standard, (viii) 38 liter TKS tank is fitted, and (ix) Introduction of Hawker 800XP designation.

Engines

2 Allied Signal Engines TFE 731-5BR turbofan engines.

Fuel

Aviation Kerosene to specification Defence Standard 91-87, NATO Code F-34, Defence Standard 91-91, NATO Code F-35, ASTM .D.1655 (Jet A or Jet A-1), CAN/COGS 3.23/ (Jet A or Jet A-1), Mil-T-83133 JP8 Grade, GOST 10227-86 (TS-1, T-1 or RT.), GB 6537-94/No.3.

Aviation Wide-cut to specification Defence Standard 91-88, NATO Code F-40, ASTM D.1655 Jet B, MIL-T-5624 JP4 and JP5 Grades, CAN/COGS 3.22/ Jet B, GOST 10227-86 T-2

Engine Limits

	TFE 731-5BR with APR not operating	TFE 731-5BR with APR operating
Take-off static thrust standard day, sea level conditions (5 minute limit) lbs.	4,750	4,750
Maximum continuous static thrust, standard day, sea level conditions (unrestricted) lbs.	4,634	4,634
Maximum permissible engine rotor operating speed		
L.P. Shaft (N1)	100 % (21,000 rpm)	100 % (21,000 rpm)
H.P. Shaft (N2)	100 % (30,300 rpm)	100.8 % (30,540 rpm)

	TFE 731-5BR with APR not operating	TFE 731-5BR with APR operating
Maximum permissible interstage turbine temperature (ITT):		
Take-off (5 minutes maximum)	978°C	996°C
Take-off (5 second maximum)	1006°C	1006°C
Take-off (2 second maximum)	1016°C	1016°C
Maximum continuous	968°C	968°C
Engine starting and relighting (unrestricted)	978°C	978°C
Engine starting and relighting (10 seconds)	996°C	996°C
Engine starting and relighting (5 seconds)	above 996°C	above 996°C
Maximum permissible oil temperature:		
Sea level to 30,000 ft.	127°C	127°C
Above 30,000 ft.	140°C	140°C
Transient temperature above maximum at any altitude for a duration of not more than two minutes	149°C	149°C
Minimum permissible oil temperature:		
Engine starting	-40°C	-40°C
Before take-off	+30°C	+30°C

Engine Limits (Cont.)

Maximum permissible air bleed extraction:

L.P. air source	5 %	5 %
H.P. air source (climb and cruise condition)	3 %	3 %
H.P. air source (descent condition only)	5 %	5 %

Airspeed Limits (IAS)

V_{MO} (maximum operating)

With fuel in the ventral tank	280 knots
With ventral tanks empty, Sea level to 12,000 ft. decreasing linearly 1 knot per 680 ft. to 310 kts. at 29,000 ft.	335 knots

M_{MO} (maximum operating) 0.80 M

M_{MO} (Mach Trimmer unserviceable/ inoperative) 0.73 M

V_A (maneuvering)

Sea level	196 knots
10,000 ft.	202 knots
20,000 ft.	207 knots
30,000 ft.	217 knots
35,000 ft.	225 knots
38,000 ft.	231 knots
40,000 ft.	236 knots
41,000 ft.	238 knots

V_{FE} (Flap speeds)

Deflection	
15^o	220 knots
25^o	175 knots
45^o	165 knots

V_{LO} (landing gear operation)

Retract	220 knots
Extend	220 knots

V_{LE} (landing gear extended) 220 knots

V_{MC} (minimum control speed)

V_{MCA} (with flaps 0^o at sea level for temperatures below 23^o C)	114.0 knots
V_{MCA} (with flaps 15^0 at sea level for temperatures below 23^0 C)	108.0 knots
V_{MCG} (with flaps 0^o or 15^O at sea level for temperatures below 23^oC)	115.5 knots
V_{MCL} (with flaps 25^o at sea level for temperatures below 23^oC)	106.0 knots
V_{MCL} (with flaps 45^o at sea level for temperatures below 23^oC)	105.0 knots

Datum

The center of gravity datum (station 353.04 inches) is 11 feet forward of the fuselage reference point. The reference point is defined by an eye bolt on the fuselage skin located beneath the starboard engine pod.

Standard Mean Chord (SMC)

87.16 in. The leading edge of the SMC is 15.70 in. forward of the datum (for SMC definition, see Approved Flight Manual).

C.G. Range (Gear and Flaps Retracted)

Wt. Lbs.	Fwd. of Datum % SMC	Fwd. of Datum In.	Aft of Datum % SMC	Aft of Datum In.
28,000***	24.80	(5.92)*	32.10	12.28
27,100***	23.40	(4.72)*	35.00	14.81
27,000***	25.60	(6.61)*	-	-
26,950***	19.40	(1.21)*	-	-
26,000	17.60	.36	-	-
25,500***	-	-	29.00	9.58
25,000	16.60	1.23	-	-
24,000***	-	-	35.00	14.81
24,000	15.70	2.02	-	-
23,000	15.25	2.45	-	-
22,600***	-	-	33.00	13.06
22,400***	-	-	28.40	9.05
22,000	15.00	2.63	-	-
21,400	-	-	26.30	7.22
20,400	-	-	26.60	7.48
18,450**	15.70	2.02	24.70	5.83
17,700	-	-	29.20	9.75
17,000**	-	-	29.60	10.10
15,750	15.00	2.63	-	-
15,465**	15.70	2.02	-	-
14,120**	19.00	(0.86)*	28.70	9.31

*(Dimension Aft of Datum)
**(Boundary Area for Zero Fuel Weight)
***(Boundary Area obtainable with Full Ventral Tank fuel)

C.G. Range (Cont.)

Item (Extending)	Moment Change In. Lb.
Wing flaps 15^{o}	+538
25^{o}	+879
45^{o}	+1593
Main landing gear	-1980
Nose landing gear	+1380

The airplane is normally weighed with wing flaps retracted.

Leveling Means

Fore and aft alignment bolts are situated in the fuselage seat rails at stations 309.35 and 371.55

Maximum Weights

Maximum ramp weight	28,120 Lbs.
Maximum brake release weight	28,000 Lbs.
Maximum landing weight	23,350 Lbs.
Maximum zero fuel weight	18,450 Lbs.
Minimum zero fuel weight	14,120 Lbs.

Minimum Crew

For all flights, 2 pilots

Maximum Passengers

15

Maximum Baggage

Compartment	Body Station	Maximum Load Lb/Ft^2	Capacity Pounds (See NOTE 8)
Forward	180.25 to 234.24	100	225
Forward cabin			
(a) Side floor	245.85 to 303.85	50	
(b) Center floor	245.86 to 303.85	60	
Aft cabin			
(a) Side floor	303.85 to 395.3	50	
(b) Center floor	303.85 to 395.3	60	
Aft	397.80 to 422.30	100	210

Fuel Capacity

Usable Fuel

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	634.00	4,223	8.20
Tank 2	634.00	4,223	8.20
Ventral tank (See NOTE 62)	233.00	1,552	100.40
Total	1,501.00	9,998	22.51

Usable Fuel (Pressure refueled)

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.
Tank 1	631.60	4,207	8.20
Tank 2	631.60	4,207	8.20
Ventral tank (See NOTE 62)	229.40	1,528	100.40
Total	1,492.60	9,942	22.61

Oil Capacity

Engine Tank Oil is the oil that is required for circulation in the system.

Location	Volume U.S. Gal	Maximum Weight Lbs.	Arm In.	Moment In. Lbs.
No. 1	1.65	12.4	90.84	1,126
No. 2	1.65	12.4	90.84	1,126
Total	3.30	24.8	90.84	2,252

Maximum Operating Altitude 41,000 feet.

Serial Numbers Eligible

258266, 258277 through 258287, 258289 through 258304, 258307 through 258324, 258326 through 258332, 258334 through 258340, 258342 through 258347, 258349 through 258359, 258361 through 258369, 258371 through 258380, 258382 through 258406, 258408 through 258426, 258428 through 258444, 258446 through 258468, 258470 through 258492, 258494 through 258512, 258514 through 258532, 258534 and on.

Data Pertinent to all Models.

Required Equipment.

The basic required equipment as prescribed in the applicable Airworthiness (See Certification Basis) and Operating Regulations must be installed in the aircraft for certification.

The BH/DH/HS/BAe.125 and Hawker Aircraft Maintenance Schedule (MS) publications reference MS.125-1/400 (Series 1A, 1B, 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3B, 3A/R, 3B/R, 3A/RA, 3B/RA, 3B/RB, 3B/RC, F3B, F3B/RA, 400A, 400B, 401B, 403A(C), 403B, F400B and F403B), MS.125-600 (Series 600A, 600B, F600B, 600B/1, 600B/2 and 600B/3), MS.125-700 (Series 700A and 700B), MS.125-800 (Series 800A, 800B and Hawker 800), MS.Hawker 800 C29A

(Hawker 800 C29A), MS U125 (Hawker 800 U-125), MS U125A (Hawker 800 U-125A), MS.800XP (Hawker 800XP) and MS.125-1000 (Series 1000A, 1000B and Hawker 1000) contain lists of all required inspection schedules pertinent to the model variants specified herein and optional equipment installations approved by the FAA, and identifies all life-limited items (See NOTE 3). Document 25.6PF.61 Fin Tank Refuel instructions is required for HS.125 Series 600A and 600B Certification. Document 25.7PF.83 Single Point Pressure Refuel Instructions is required for HS.125 Series 700A and 700B Certification. Document 28.8PF.59-1 Pressure Refueling Instructions is required for BAe.125 Series 800A, 800B and Hawker 800 Certification. Document 25-9PF 212 Pressure Refueling Instructions is required for BAe.125 Series 1000A, 1000B and Hawker 1000 Certification.

Control Surface Movements.

To ensure proper operation of the airplane the movement of the various control surfaces must be carefully controlled by proper rigging of the flight control systems. The airplane must, therefore, be rigged according to the approved data contained in the Maintenance Manuals (MM or AMM). Publication reference MM.125 (Series 1A, 1B, 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3B, 3A/R, , 3B/R, 3A/RA, 3B/RA, 3B/RB, 3B/RC, F3B, F3B/RA, 400A, 400B, 400B/1, 401B, 403A(C), 403B, F400B, F403B, 600A, 600B, 600B/1, 600B/2, 600B/3 and F600B) or MM.125-700 (HS.125 Series 700A and 700B), AMM 125-800 Vol. 1-3 (BAe.125 Series 800A, 800B and Hawker 800), AMM C29A Vol. 1-3 (Hawker 800 C29A), AMM U125 Vol. 1-3 (Hawker 800 U-125), AMM U125A Vol. 1-3 (Hawker 800 U-125A), AMM 125-1000A Vol. 1-3 (BAe.125 Series 1000A, 1000B and Hawker 1000) or AMM Hawker 800XP Vol. 1-3 (Hawker 800XP).

Certification Basis.

Application for Type Certificate to the UK CAA was dated September 28, 1960. CAR.10, British Civil Airworthiness Requirements (1st November 1963), and Special Conditions notified by the United States Government to the Government of the United Kingdom including Validation Arrangements (V.A.) Note 1, Issue 1 dated April 19, 1961. This certification is equivalent to CAR.4b dated December 1953, Amendment 4b-1 through 4b-11, exclusive of CAR 4b.350 (e) and includes Special Regulation SR.422B.

CAR.10, Type Certificate No. A3EU issued September 25, 1964. The Type Certificate was amended February 3, 1966, to include Model DH.125 Series 1A-522, amended November 7, 1966 to include Model DH.125 Series 3A, amended August 9, 1967 to include Model DH.125 Series 1A/R-522 and Model DH.125 Series 3A/R, amended February 15, 1968, to include Model DH.125 Series 1A/S-522 and Model DH.125 Series 3A/RA, amended November 15, 1968, to include Model DH.125 Series 400A, amended July 14, 1970, to include BH.125 Series 400A, amended August 17, 1972, to include BH.125 Series 600A, amended January 6, 1976, to include HS.125 Series 600A, amended May 20, 1977, to include HS.125 Series 700A, amended July 12, 1984 to include BAe.125 Series 800A, amended October 31, 1991, to include BAe.125 Series 1000A, amended January 28, 1994, to include Hawker 800 and Hawker 1000, amended July 28, 1995, to include Hawker 800XP, amended May 28, 1999 to include HS.125 Series 1B, 1B-522, 1B/R-522, 1B/S-522, 3B, 3B/R, 3B/RA, 3B/RB, 3B/RC, F3B, F3B/RA, 400B, 400B/1, 401B, 403A(C), 403B, F400B, F403B, 600B, 600B/1, 600B/2, 600B/3, F600B, 700B, and BAe.125 Series 800B and 1000B.

Compliance, over and above, certification basis requirements, has been met with CAR Amendments 4B-12 and 4B-14. Compliance has been established with the following optional requirements: Ice Protection Provision 4b.640. FAA Exemption No. 573 grants exemption from CAR 4b.437, however for DH.125 Series 400A and subsequent models added to this Type Certificate, compliance has been established for Fuel Jettisoning Systems certification weights with FAR 25.1001 of Amendment 25-18. For BH.125 and HS.125 Series 600A models compliance has been established with the special retroactive requirements of FAR 25.2 through FAR Amendment 25-20 and FAR 21 Amendment 21-27 and (FAR 36(1)(c)(2)). (See NOTE 18).

HS.125 Series 1A, 3A, 3A/RA and BH/HS.125 Series 400A and 600A airplanes fitted with Garrett AiResearch TFE 731-3 engines comply with the later requirements of FAR 21.183(e) amendment 21-42 and FAR 36.1(d) amendment 36-1 through 36-5.

For HS.125 Series 700A models, compliance has been established with the special retroactive requirements of FAR 25.2 through amendment 25-20; FAR 25.979 of amendment 25-11, FAR 21.183(e) of amendment 21-42, and FAR 36.1(d) for amendment 36-1 through 36-5.

Certification Basis. (Cont.)

For BAe.125 Series 800A models, compliance has been established with the specific additional requirements of FAR Part 25, Amendment 25-1 through 25-54, above and beyond the CAR. 10, British Civil Airworthiness Requirements specified in the second paragraph above under "Certification Basis." The additional FAR requirements are as follows:

FAR 25.2
FAR 25.305 (For wing only)
FAR 25.571 (For wing and engine mounts only) (See NOTE 3)
FAR 25.903 (d) (1)
FAR 25.979 (a) through (c)
FAR 25.1419
FAR 25.1529

Plus FAR Part 36 as amended by Amendment 36-1 through 36-12.
Plus Special Federal Aviation Regulation (SFAR) 27 as amended by Amendments 27-1 through 27-4.

Equivalent Safety is established with:

FAR 25.773(b)(2) - Pilots Window
FAR 25.613(a) - Design Values
FAR 25.615(a) - Design Properties

For Hawker 800XP airplanes fitted with Allied Signal TFE 731-5BR engines:
The U.S. Certification Basis for BAe.125 Series 800A models (including equivalent safety findings) and, in agreement with the manufacturer, compliance has been established with the following additional FAR requirements:

For the Engine Electronic Controls and Mach Trim System:

FAR 25.1316 as amended through amendment 25-80 and
Special Condition No. 25-ANM-98 High-Intensity Radiated Fields

For the thrust reverser installation:

FAR 25.933 as amended through amendment 25-40
FAR 25.934 as amended through amendment 25-23
FAR 25.1309 as amended through amendment 25-23

Plus FAR Part 34 basic issue.

Plus FAR Part 36 as amended by Amendments 36-1 through 36-20

For BAe.125 Series 1000A models: The U.S. Certification Basis for BAe.125-800A models (including equivalent safety findings) and, in agreement with the manufacturer, compliance has been established with specific additional requirements of Part 25 of the FAR, as amended by amendments 25-1 through 25-70, for areas of significant design change from the Series 800A. The additional FAR requirements are as follows:

25.25	25.1021
25.33	25.1045(d)
25.361(b)	25.109(e)
25.365(a) and (d)	25.1093(b)(1)(i)(ii)
25.511(b)(6)	and (b)(2) (Engine only)
25.571(b)(6)	25.1141(f)(2)
25.697(a)	25.1143(d)
25.735(f)(1)	25.1163(a)
25.843(a)	25.1183(b)(1)
25.853(b) and (c)	25.1189(a)(1) and (2)
25.855(a)	25.1303(c)(1)
25.857(d)(6)	25.1305(c)(6) and (7)
25.901(c)	25.1309(a), (b), (c), (d) and (e)
25.903(a)	25.1323(b)(2)
25.904	25.1331(a)(3)
25.905	25.1359
25.939(a)	25.1411(a)
25.961	25.1423
25.963(e)	25.1438(a)(b) and (c)
25.993(c)	25.1457(c)
25.994	25.1459(a)(4) and (e)
25.997	25.1521(b) and (c)
25.1001	25.1549 (Engine only)
25.1013	
25.1015	Appendix F
25.1019	Appendix H

NOTE: Compliance with the subject paragraphs of FAR 25.1309 has been established for systems which have been significantly redesigned.

Plus FAR Part 36 as amended by Amendments 36-1 through 36-18.

Plus Special Federal Aviation Regulation (SFAR) 27 as amended by Amendments 27-1 through 27-6.

Plus Special Conditions: Special Conditions No. 25-ANM-34 dated June 29, 1990, High Altitude Operation and Protection from Effects of Lightning and High Intensity Radiated Fields.

The BH/DH/HS/BAe.125 Series (1B, 1B-522, 1B/R-522, 1B/S-522, 3B, 3B/R, 3B/RA, 3B/RB, 3B/RC, F3B, F3B/RA, 400B, 400B/1, 401B, 403A(C), 403B, F400B, F403B, 600B, 600B/1, 600B/2, 600B/3, F600B, 700B, 800B and 1000B) and some Hawker 800 and 1000 models were certified to CAA, United Kingdom, regulations. As of May 28, 1999, these 'B' aircraft are eligible to receive FAA Airworthiness Certificates and Registration as a 'B' aircraft if shown to meet the requirements to be equivalent to an 'A' aircraft.

Production Basis

TC only: Serial numbers 258297, 258301, 258304, 258306 and 259003. Prior to Standard Airworthiness, Aircraft must be inspected and flight tested by FAA.
Production Certificate, PC-8: Serial numbers 258309, 258311, 258313, 258315, 258317, 258319, 258320, 258322, 258325, 258326, 258331, 258333, 258334, 258336 and 258338 and on.

Service Information. Service bulletins, structural repair manuals, repair drawings, vendor manuals, aircraft flight manuals, and overhaul and maintenance manuals, which contain a statement that the document is C.A.A. approved, or C.A.A. approved through the Manufacturer's C.A.A. Approval Reference, DAI/1103/38, DAI/1011/55 or DAI/2652/55, prior to August 1, 1995, are accepted by the FAA and are considered FAA approved. These approvals pertain to the type design only. Effective August 1, 1995 and after, service information pertaining to the type design is to be FAA approved under FAR Part 21 requirements.

NOTES.

NOTE 1. (a) A current weight and balance report, including list of equipment in certificated empty weight and loading instructions, must be provided for each aircraft at the time of original certification.
(b) The airplane must be loaded so that the C.G. is within the specified limits at all times with the effect of fuel use and movement of crew and passengers from their assigned positions being considered.
(c) The "drainable unusable fuel" is the amount of fuel in the tanks which is unavailable to the engines under critical flight conditions as defined in CAR 4b.416. This drainable unusable fuel does not include the "tank trapped fuel". The total unusable fuel must be included in the airplane empty weight or be suitably accounted for in the airplane weight and balance report. The total volume of unusable fuel in gallons is as follows:

Airplane Total. (BH/DH/HS/.125 Series 1A, 1B, 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3B, 3A/R, 3B/R, 3A/RA, 3B/RA, F3B/RA, 400A, 400B, 400B/1, F400B, 401B, 403A(C), 403B, and F403B)

	Volume U.S. Gal	Weight Lbs.	Arm In.
Tank trapped	2.4	16	12
Drainable unusable fuel	9.0	60	16.5
Total unusable fuel	11.4	76	15.6

Airplane Total. (BH/HS.125 Series 600A, 600B, 600B/1, 600B/2, 600B/3, F600B and HS.125 Series 700A and 700B).

	Volume U.S. Gal	Weight Lbs.	Arm In.
Tank trapped	3.4	11.6	-16.6
Drainable (Wing)	11.5	76.6	-14.0
Unusable (Ventral)	0.9	6.0	59.0
Fuel (Dorsal)	Nil	Nil	-
Total unusable fuel	15.8	105.2	-9.7

Airplane Total. (BAe.125 Series 800A, 800B, Hawker 800 and Hawker 800XP)

	Volume U.S. Gal	Weight Lbs.	Arm In.
Tank trapped	3.3	22.0	-15.6
Drainable (Wing)	8.1	54.0	-14.0
Unusable (Ventral)	0.7	5.0	74.4
Total unusable fuel	12.1	81.0	9.0

Airplane Total. (BAe.125 Series 1000A and 1000B and Hawker 1000)

	Volume U.S. Gal	Weight Lbs.	Arm In.
Tank trapped	3.3	22.0	-15.6
Drainable (Wing)	8.1	54.0	-14.0
Unusable forward ventral	4.2	28.0	-57.6
Unusable aft ventral	2.6	17.6	60.0
Total unusable fuel	18.2	121.6	-13.1

(d) Engine System oil is the total engine oil less than the quantity drainable from the tank. The undrainable oil for the following aircraft fitted with Viper engines is:
(BH/DH/HS/.125 Series 1A, 1B, 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3B, 3A/R, 3B/R, 3A/RA, 3B/RA, 400A, 400B, 400B/1, 401B, 403A(C), 403B, 600A, 600B, 600B/1, 600B/2 and 600B/3 fitted with Viper Engines).

Location	Volume U.S. Gal.	Weight Lb.	Arm In.	Moment In. Lb.
No. 1	0.3	2.3	82.5	190.4
No. 2	0.3	2.3	82.5	190.4
Total	0.6	4.6	82.5	380.8

The undrainable oil for the following aircraft fitted with Garrett TFE 731 engines is:
(BH/DH/HS/BAe.125 Series 1A, 1B, 3A, 3B, F3B/RA , 400A, 400B, F400B, F400B, 401B, 403A(C), 403B, F403B, 600A, 600B, 600B/1, 600B/2, 600B/3, F600B, 700A, 700B, 800A, 800B, Hawker 800 and Hawker 800XP fitted with Garrett TFE 731 engines).

Location	Volume U.S. Gal.	Weight Lb.	Arm In.	Moment In. Lb.
No. 1	1.5	11.3	106.2	1200
No. 2	1.5	11.3	106.2	1200
Total	3.0	22.6	106.2	2400

(BAe.125 Series 1000A and 1000B and Hawker 1000A and 1000B fitted with Pratt & Whitney PW305B engines.)
The total quantity of oil for both engines is 6.02 U.S. gallons. The weight of this is included in the Basic Aircraft Weight.

NOTE 2. Any 'A' and 'B'(operating as 'A' equivalent) aircraft must be operated according to the appropriate FAA Approved Flight Manual, Document No. HS.1.2 (DH.125 Series 1A and HS.125 Series 1B) or Document No. HS.1.3 (DH.125 Series 1A-522, 1A/R-522, 1A/S-522, 3A, 3A/R and 3A/RA) or Document No. HS.1.3 (HS.125 Series 1B-522, 1B/R-522. 1B/S-522, 3B, 3B/R, 3B/RA, 3B/RB and 3B/RC) or Document No. HS.1.5 (DH/BH.125 Series 400A and HS.125 Series 400B, 401B, 403A(C) and 403B), or Document No. HS.1.7 (BH.125 Series 600A and HS.125 Series 600A, 600B, 600B/1, 600B/2 and 600B/3) or Document No. HS.1.9 (BH/HS.125 Series 600A with Modification 252468, HS.125 Series F600B, 700A and 700B) or Document No. HS.1.10 (DH.125 Series 3A/RA with Modification 252600, DH/BH.125 Series 400A with Modification 252550, HS.125 Series F3B/RA, F400B and F403B), or Document No. HS.1.11 (DH.125 Series 1A with Modifications 251867 and 252605, DH.125 Series 1A with Modification 252606, and DH.125 Series 3A with Modification 252603 and HS.125 Series F3B) or Document No. H.S.1.16 (BAe.125 Series 800A and 800B and Hawker 800), or Document No. HS.1.19 (BAe.125 Series 1000A and 1000B and Hawker 1000) or Document No. HS.1.22 (Hawker 800XP).

As of August 1, 1995, the FAA accepted responsibility for the maintenance and approval of all Airplane Flight Manuals incorporated by reference within this data sheet and those manuals and amendments thereof previously issued by the United Kingdom Civil Aviation Authority in association with DH/HS/BH/BAe.125 Series 1 through 1000 and Hawker 800, 800XP and 1000 Series products designed and/or manufactured under its authority. All such manuals must incorporate the following amendments which relate to this transfer of responsibility.

AFM No.	Particular Amendment	Model Applicability
HS.1.2	P 25	Series 1A/B Models
HS.1.3	P 91	Series 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3B, 3A/R, 3B/R, 3A/RA, 3B/RA, 3B/RB, and 3B/RC Models
HS.1.5	P 44	Series 400A, 400B, 400B/1, 401B, 403A(C) and 403BModels
HS.1.7	P 37	Series 600A, 600B, 600B/1, 600B/2 and 600B/3 Models
HS.1.8	P 47	UK CAA Compliant Series F600B and 700B Models
HS.1.9	P 41	US FAA Compliant Series 700A & Garrett converted Series 600A Models
HS.1.10	P 17	Garrett converted Series 3A/RA (with Long Range fuel), F3B/RA, 400A, F400B, and F403B Models
HS.1.11	P 9	Garrett converted Series 1A and 3A (without Long Range fuel) Models
HS.1.15	P 57	UK CAA Compliant Series 800 and Hawker 800 Models
HS.1.16	P 70	US FAA Compliant Series 800 and Hawker 800 Models
HS.1.18	P 40	UK CAA Compliant Series 1000 and Hawker 1000 Models
HS.1.19	P 40	US FAA Compliant Series 1000 and Hawker 1000 Models
HS.1.22	*	US FAA Compliant Hawker 800XP Models
149-590032-0005	*	US FAA Compliant Hawker 800XP Models

* Original Manual issued in the U.S.

NOTE 3. Service Life Limits and Airworthiness Structural Inspections:

Service Life Limits of Structural Components. The service life limits for aircraft structural parts which are fatigue critical are listed in Raytheon Aircraft Company Document Reference CJE-HPA-C-GEN -AW1667, latest FAA approved revision.

Airworthiness Structural Inspections. For the BAe.125 Series 800/Hawker 800/Hawker 800XP the structural inspections specified in CJE.HTS.R.258.AW0949, Issue 1 are essential to ensure the continued airworthiness of the BAe.125 Series 800/Hawker 800/Hawker 800XP in operational service. The inspections may be changed only with the mutual agreement between the airworthiness authorities, the aircraft manufacturer and the operator.

NOTE 4. Kerosene type and wide-cut type fuels conforming to the specifications in the data sheet may be used separately or mixed in any proportions. When the fuel type has been changed, a check must be made at the subsequent take-off to confirm that either the appropriate maximum r.p.m. or maximum ITT is being achieved. Aviation gasoline meeting the following specifications may be used within the limits specified in the appropriate Approved Flight Manual or Supplement:

American: Mil-G-5572, JP4 and JP5 Grades; MIL-T-83133 JP8 Grade, ASTM D1655/JET A, JET A-1 and JET B Grades.
British Defense Standards: 91-87, 91-90 and 91-91.
Canadian: 3-GP-23h; 3-GP-25; CAN/CGSB 3.23/Jet A and Jet A-1; CAN/CGSB 3.22/Jet B
Russian: GOST 10227-86/ T-1, T-2, TS-1 premium and RT Grades.
Chinese: GB 6537-94/ No. 3

NOTE 5. Airworthiness Certification for aircraft manufactured in the United Kingdom prior to August 1, 1995, and delivered new to the United States. An acceptable minimum standard of equipment was installed on production DH.125 Series aircraft for factory flyaway (ferrying) on a United Kingdom Certificate of Airworthiness for Export. This standard was in accordance with Parts 2 and 6 (and related Appendices and Addendum) of Airworthiness document DO/AW/125/FAA/TC.1 current issue (DH.125 Series 1A, 1A-522, 3A, 1A/R-522, 3A/R, 1A/S-522, 3A/RA, 400A) or DO/AW/125-600/FAA/TC.1 current issue (BH.125 Series 600A, HS.125 Series 600A) or DO/AW/125-700/FAA/TC.1 current issue (HS.125 Series 700A) or DO/AW/125-800/FAA/TC.1 current issue (BAe.125 Series 800A/Hawker 800) or CJE.HPA.C.258.AW2017 current issue (Hawker 800XP) or Parts 2 and 5 (and related Appendices and Addendum) of Airworthiness document CJE-HPA-C-260-AW1660 current issue (BAe.125 Series 1000A and Hawker 1000).

A Standard U.S. Certificate of Airworthiness was issued on proof of satisfactory conformance with . Modifications listed in Part 5 (and related Appendices and Addendum) of the above referenced documents except Part 4 of document CJE-HPA-C-260-AW1660 (BAe.125 Series 1000A and Hawker 1000), current issue. Current issues of Documents DO/AW/125/FAA/TC.1, DO/AW/125-600/FAA/TC.1, DO/AW/125-700/FAA/TC.1, DO/AW/125-800/FAA/TC.1, CJE-HPA-C-258-AW2017 and CJE-HPA-C-260-AW1660 may be obtained upon request to the manufacturer.

Each individual aircraft at delivery was further identified as to status of incorporation of factory-installed modifications by the "Modification Statement" appended to the aircraft logbook.

NOTE 6. Airworthiness Certification for aircraft manufactured in the United Kingdom after August 1, 1995

FAA Standard Airworthiness Certificates and Export Certificates of Airworthiness may be issued to aircraft manufactured in the UK by Raytheon Corporate Jets, Inc. under license from Raytheon Aircraft Company after August 1, 1995, based on the following:

a. Exemption Number 6142 granted to Raytheon Aircraft Company on August 3, 1995, from FAR 21.183(c) and FAR 21.325(b)(1) for Hawker 800, 800XP and 1000 aircraft.

b. A certifying statement from the UK CAA stating the aircraft has been examined, tested and found to conform to US Type Certificate A3EU and is in a condition for safe operation.

c. The aircraft must be fitted with data plates conforming to FAR 45.13 and stating that Raytheon Corporate Jets, Inc. is the builder under license from Raytheon Aircraft Company.

d. The following serial numbered aircraft were manufactured in the UK by Raytheon Corporate Jets, Inc. under license to Raytheon Aircraft Company.

Hawker 800 (U-125A) 258268, 258288 and 258305
Hawker 800XP 258266, 258277 through 258287, 258289 through 258296, 258298 through 258300, 258302, 258303, 258307, 258308, 258310, 258312, 258314, 258316, 258318, 258321, 258323, 258324, 258327 through 258330, 258332, 258335, 258337.
Hawker 1000 259048 through 259052.

NOTE 7. Maximum permissible turbine outlet gas temperatures with Modification 251760 embodied are:

Takeoff (5 minutes maximum)	740°C
Maximum continuous	715°C
Maximum for acceleration	715°C
Starting maximum gas temp.	800°C

NOTE 8.

Maximum Cabin Loads	Total	Forward of Front Spar Frame Datum	Aft of Front Spar Frame Datum
DH/HS/BH.125 Series -1A, 1A with modification 252605, 1A with modifications 251867 and 252606, 1B, 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3A with modification 252603, 3B, 3A/R, 3B/R, 3A/RA, 3B/RA, 400A, 400B, and 400B/1	1950 lbs.	900 lbs.	1350 lbs.
HS.125 Series 401B	2250 lbs.	1020 lbs.	1350 lbs.
HS.125 Series 403A(C) and 403B DH.125 Series 3A/RA with modification 252600 DH/BH 125 Series 400A, HS.125 Series F3B/RA, F400B and F403B	2300 lbs.	1150 lbs.	1350 lbs.

Maximum Cabin Loads (Cont.)	Total	Forward of Front Spar Frame Datum	Aft of Front Spar Frame Datum
BH/HS.125-600A and HS.125 Series 600B	2960 lbs.	1515 lbs.	1670 lbs.
BH/HS 125 Series 600A and 600B(with modification 252320), BH/HS 125-600A with modification 252468), HS.125 Series F600B, HS.125 Series 700A, and 700B, BAe.125 Series 800A and 800B, Hawker 800 and Hawker 800XP	3050 Lbs.	1550 Lbs.	1680 Lbs.
BAe.125 Series 1000A, 1000B and Hawker 1000	3070 lbs.	1660 lbs.	1910 lbs.

Maximum load forward or aft of front spar frame datum includes loads of passenger seats, their occupants (including supernumerary) and of the appropriate luggage compartments.

To ensure that airplane C.G. is within allowable limits it may be necessary to reduce loads to less than maximum stated above.

NOTE 9. The maximum permissible altitude is 40,000 feet except that the maximum permissible altitude can be 41,000 feet when equipment installations are incorporated conforming with either:

1. U.S. Federal Aviation Administration Supplemental Type Certificates SA858WE, SA859WE, and SA860WE and suitably modified to maintain a cabin pressure equivalent to an altitude of 8,000 ft. or;
2. Modifications No. 251600 plus 251601, or 251721, or 252210 plus 252260 and 252261A.

NOTE 10. Conversion of DH/HS.125 Series 1A-522, 1B-522, 3A or 3B respectively to 1A/R-522, 1B/R-522, 3A/R or 3B/R , may only be accomplished by incorporation of Approved Service Bulletins corresponding to Modifications No. 251700, 255640 and 255718.

NOTE 11. Conversion of DH.125 Series 1A-522 or HS.125 Series 1B-522 respectively to Series 1A/S-522 or Series 1B/S-522 may only be accomplished by incorporation of Approved Service Bulletin corresponding to Modification No. 251867. Conversion of DH.125 Series 3A/R or HS.125 Series 3B/R respectively to Series 3A/RA or Series 3B/RA may only be accomplished by incorporation of Approved Service Bulletin corresponding to Modification No. 251916.

NOTE 12. The Maximum Ramp Weight and Maximum Zero Fuel Weight for the Model DH/HS 125 Series 1A, Series 1B, Series 1A/522, Series 1B/522, Series 1A/S-522, Series 1B/S-522, Series 3A and Series 3B may be increased 200 pounds provided the revised limitation placard is installed in accordance with Modification No. 252022 and the relevant Approved Flight Manual revision is used.

NOTE 13. When engine anti-icing is in use the maximum permissible oil inlet temperature for continuous operation is 135^{O}C. This value may only be used, when Modification 252149, introducing a modified oil temperature gauge and engine limitation placard, is embodied or an approved equivalent standard is achieved.

NOTE 14. The Model DH.125 Series 1A and HS.125 Series 1B can be converted retrospectively to a Series 1A-522 and Series 1B-522 by the introduction of Modifications 251301, 251665, 251392, 251591, 251642, 251658, 251659, 257104, 255567 and 251760 in accordance with Service Bulletin 71-9-1301. The Approved Flight Manual, Document No. HS.1.2 (DH.125 Series 1A or HS.125 Series 1B) must be returned to RAC and the Approved Flight Manual, Document HS.1.3 (DH.125 Series 1A-522 or HS.125 Series 1B) must be obtained.

NOTE 15. The limiting Mach Number is reduced to 0.755 when a Smiths combined ASI/Mach meter Part No. PW 202AMA/4, PW 202AMA/6 or PW 202AMA/8 is fitted. These instruments incorporate a mach scale corrected for position error.

NOTE 16. The maximum ramp weight and maximum Zero Fuel Weight for the model DH/HS/BH 125 Series 400A and Series 400B may be increased 300 lbs., or 500 lbs. provided V_{MO} is appropriately reduced.

Modification 252243 Part C and the related Approved Flight Manual HS.1.5 and Particular Amendment No. P 14 must be embodied to permit the 300 lb. increase for a maximum zero fuel weight and a maximum ramp weight of 14,500 lbs. and 23,600 lbs. respectively

Modification 256403 Part D and the related Approved Flight Manual HS.1.5 and Particular Amendment No. P 17 must be embodied to permit the 500 lb. increase for a maximum zero fuel weight and a maximum ramp weight of 14,700 lbs. and 23,800 lbs. Respectively.

NOTE 17. (a) The maximum operating speeds (V_{MO}/M_{MO}) for the BH.125 Series 600A, HS.125 Series 600A and HS.125 Series 600B may be increased by embodiment of Modification 252320 Part A. Particular Amendment No. P 8 must be incorporated in the Approved Flight Manual Document No. HS.1.7 to permit operation at the increased speeds.

(b) The maximum ramp weight and the maximum fuel weight for the BH.125 Series 600A, HS.125 Series 600A and HS.125 Series 600B may be increased by the embodiment of Modification 252320 Part C. Particular Amendment No. P 9 must be incorporated in the Approved Flight Manual Document No. HS.1.7 to permit operation at the increased weights defined in this sub-paragraph.

(c) The maximum brake release weight for the BH.125 Series 600A, HS.125 Series 600A and HS.125 Series 600B may be increased by the embodiment of Modification 252320 Part D. Particular Amendment No. P 10 must be incorporated in the Approved Flight Manual Document No. HS 1.7 to permit operation at the increased weights defined in this sub-paragraph.

NOTE 18. In addition to the requirements listed under "Certification Basis", a BH.125 Series 600A, HS.125 Series 600A and HS.125 Series 600B airplane which has accumulated no flight time by December 31, 1974, must comply with FAR.21.183 and FAR.36.1(d)(3) Amendment 36-2 in order to qualify for the issue of a U.S. Standard Airworthiness Certificate. Compliance may be accomplished by incorporation of Modification No. 252405 and 252384. Particular Amendment No. P 18 and Supplement No. 12 must be incorporated in the associated Approved Flight Manual Document No. HS.1.7 when the above modifications are fitted.

NOTE 19. The Rolls Royce Viper engines originally fitted to BH/HS 125 Series 600A and the HS.125 Series 600B airplanes may be replaced by Garrett AiResearch TFE 731-3 Turbofan engines by embodiment of modification 252468 or equivalent and the complementary modifications listed therein. In addition to the installation of the TFE 731-3 engines, the above modification also introduces changes to systems consequential to the engine change and a reduction in takeoff (brake release) weights and increase in Maximum zero taxi weight. A BH/HS 125 Series 600A airplane modified as specified above is to be operated in accordance with the Approved Flight Manual, Document No. HS.1.9 with Particular Amendment No. P 7. The HS.125 Series F600B aircraft modified as specified herein and meeting the requirements of Note 54 must also be operated in accordance with the Approved Flight Manual Document HS 1.9 with Particular Amendment No. P 7.

NOTE 20. Modifications 252622 (Parts A and B) and 258169 (HS.125 Series 700A and Series 700B) or relevant part of 258469 (All Series except Series 700) introduce an Automatic Performance Reserve (APR) system. When these modifications are embodied, the designation of the Garrett AiResearch TFE 731-3 engine must be changed to TFE 731-3R. Limitations and procedures associated with the APR system are provided in the Approved Flight Manuals, Document H.S.1.9 Particular Amendment No. P11 (BH/HS 125 Series 600A with modification 252468, HS.125 Series F600B, 700A, and 700B.), Document H.S.1.10 with Particular Amendment No. P 2 (DH 125 Series 3A/RA with modification 252600, HS.125 Series F3B/RA, BH/DH 125 Series 400A with modification 252550 and HS.125 series F400B) and Document H.S.1.11 with Particular Amendment No. P 4 (DH.125 Series 1A with modifications 251867 and 252605, DH.125 Series 1A with modification 252606, DH.125 Series 3A with modification 252603 and HS.125 Series F3B).

NOTE 21. The Rolls-Royce Viper engines originally fitted to DH/BH.125 Series 400A airplanes, may be replaced by Garrett AiResearch TFE 731-3 turbofan engines by embodiment of Modification 252550 and the complementary modifications listed therein. The HS.125 Series 400B airplanes may replace the Garrett AiResearch TFE 731-3 turbofan engines by embodiment of Modification 252551 and the complementary modifications listed therein. In addition to the installation of the TFE 731-3 engines, the above modifications also introduce changes to systems consequential to the engine change and an increase in certificated taxi and take-off (brake release) weights. The DH/BH.125 Series 400A aircraft modified as specified herein must be operated in accordance with the Approved Flight Manual Document HS.1.10. The HS.125 Series F400B or F403B aircraft modified as specified herein and meeting the requirements of Note 53 must also be operated in accordance with the Approved Flight Manual Document HS 1.10.

NOTE 22. The Rolls-Royce Viper engines originally fitted to DH.125 Series 1A and HS.125 Series 1B airplanes may be replaced by Garrett AiResearch TFE 731-3 turbofan engines embodiment of Modification 252605 (aircraft fitted with Modification 251867) or 252606 (aircraft not fitted with Modification 251867) and the complementary modifications listed therein. In addition to the installation of the TFE 731-3 engines, the above modifications also introduce changes to systems consequential to the engine change. A DH.125 Series 1A or Series 1B aircraft modified with Modifications 251867 and 252605 as specified herein must be operated in accordance with the Approved Flight Manual Document H.S.1.11 basic.

A DH.125 Series 1A or Series 1B aircraft modified with Modification 252606 as specified herein must be operated in accordance with the Approved Flight Manual Document H.S.1.11 and Particular Amendment No. P 2.

A DH.125 Series 1B aircraft with either modifications seeking U.S. FAA Airworthiness Certificates and registration must also meet the requirements of Note 52.

NOTE 23. Modification 252672 introduces a revised landing flap setting of 45 degrees to Series 400A and earlier Viper powered airplanes. Embodiment of this modification gives a noise reduction which meets the noise requirements of I.C.A.O. Annex 16. The limitations and procedures associated with this modification are provided in the Approved Flight Manual Documents: H.S.1.2 with Particular Amendment No. P 22 (Series 1A and 1B); H.S.1.3 with Particular Amendment No. P 87 (Series 1A-522, 1B-522, 1A/R-522, 1B/R-522, 1A/S-522, 1B/S-522, 3A, 3B, 3A/R, 3B/R, 3A/RA, 3B/RA and 3B/RB); H.S.1.5 with Particular Amendment No. P 37 (Series 400A and 400B)

NOTE 24. Modification 256991 introduces an Aeronca Thrust Reverser system to the HS.125 Series 700A and Series 700B aircraft. The limitations and procedures associated with the thrust reverser system are provided in the Approved Flight Manual Document H.S.1.9, by Particular Amendment No. P 12.

NOTE 25. The Rolls-Royce Viper engines originally fitted to DH.125 Series 3A/RA and HS.125 Series 3B/RA airplanes may be replaced by Garrett TFE 731-3 turbofan engines by embodiment of Modification 252600 or equivalent and the complementary modifications listed therein. In addition to the installation of the TFE 731-3 engines, the above modifications also introduce changes to systems consequential to the engine change and an increase in certificated taxi and take-off (brake release) weights. A DH.125 Series 3A/RA aircraft modified as specified herein must be operated in accordance with the Approved Flight Manual Document. H.S.1.10. A HS.125 Series F3B/RA aircraft modified as specified herein and meeting the requirements of Note 52 must also be operated in accordance with the Approved Flight Manual Document HS.1.10.

NOTE 26. The Rolls-Royce Viper engines originally fitted to DH.125 Series 3A and HS.125 Series 3B aircraft may be replaced by Garrett TFE 731-3 turbofan engines by embodiment of Modification 252603 on the DH.125 Series 3A and Modification 252604 on the HS.125 Series 3B and the complementary modifications listed therein. In addition to the installation of the TFE 731-3 engines, the modifications also introduce changes to systems consequential to the engine change. A DH.125 Series 3A aircraft modified as specified herein, must be operated in accordance with the Approved Flight Manual Document HS 1.11 with Particular Amendment No. P 3. A HS.125 Series F3B aircraft modified as specified herein and meeting the requirements of Note 52 must also be operated in accordance with the Approved Flight Manual Document HS 1.11 with Particular Amendment No. P 3.

NOTE 27. The maximum ramp weight may be increased by 500 lbs. to 25,500 lbs. and the take-off weight by 700 lbs. to 25,500 lbs. for the BH/HS.125 Series 600A with modification 252468, HS.125 Series F600B, HS.125

Series 700A and Series 700B aircraft providing that a revised limitations label is installed in accordance with Modification 252818 (Series 600A) or 258332, (Series 700) and the Approved Flight Manual document HS.1.9 containing Particular Amendment No. P 13.

NOTE 28. Aviation Wide-cut fuel may only be used with TFE 731 engined aircraft when both engines have Modification 252738 embodied.

NOTE 29. The maximum zero fuel weight may be increased with reductions in V_{MO} on a HS.125 Series 700A and Series 700B aircraft with modifications 252648 and 258332 by embodiment of modification 258825 Part D and by inclusion in the Approved Flight Manual Document H.S.1.9 of Particular Amendment No. P 26.

NOTE 30. Modification 259550A introduces the BAe.125 Series 800A (C-29A C-FIN aircraft) intended for operation by the United States Air Force. Document HAW.D.258.AW0159 Issue 4 outlines the changes made to the standard BAe.125 Series 800A aircraft to achieve the delivery standard exported from the manufacturer. These aircraft embody features which would not normally be found on civil aircraft, including various provisions to enable the aircraft to be completed to the USAF requirements in the USA (Where provisions have been made for the fitment of equipment by the U.S. customer, these have been shown to comply with the associated installation requirements and be of no hazard to the aircraft, but have not been investigated for their intended function.)

A BAe.125 Series 800A aircraft modified as specified above must be operated in accordance with the Approved Manual Document No. HS 1.16 containing Particular Amendment No. P 40 and any other applicable approved amendments.

NOTE 31. The maximum taxiing (ramp) weight and the maximum take-off (brake release) weight for the BAe.125 Series 800A, 800B and Hawker 800 aircraft may be increased to 28,100 lbs. (12,746 kg) and 28,000 lbs. (12,701 kg) respectively, by the embodiment of either Modification 259550 Part B or 259952 Part A.
An aircraft modified as specified above must be operated in accordance with the Approved Flight Manual Document No. HS.1.16 containing Particular Amendment No. P 45.

NOTE 32. The maximum zero fuel weight for the BAe.125 Series 800A, 800B and Hawker 800 aircraft may be increased to 18,000 lbs. when Modification 253169A is embodied.

NOTE 33. When a baggage pannier (Mod. 259292 or 259500) is embodied in lieu of the ventral tank, V_{MO} is:
335 knots up to 12,000 feet, less 1 knot per 680 feet, to 310 knots at 29,000 feet.

NOTE 34. The maximum zero fuel weight may be increased to 16,300 Lbs. for HS.125 Series 700A and Series 700B with Modification 258825 embodied.

NOTE 35. The Maximum Zero Fuel Weight may be increased to 15,200 lbs. but with a reduction in V_{MO} on a DH/BH.125 Series 400A or HS.125 Series 400B with Modification 259273 embodied.

NOTE 36. Modification 259283 introduces Dee Howard TR5000BR Thrust Reversers to the BAe.125 Series 800A, 800B and Hawker 800. The limitations and procedures associated with the thrust reverser are provided in the Approved Flight Manual Document HS.1.16 containing Particular Amendment No. P 32.

NOTE 37. The maximum zero fuel weight for the BAe.125 Series 800A, 800B and Hawker 800 aircraft may be increased to 17,750 lbs. when Modification 259579A is embodied.

NOTE 38. The maximum zero fuel weight on the BAe.125 Series 1000A, 1000B and the Hawker 1000 may be increased to 20,300 lbs. provided that a revised limitations label is installed in accordance with Modification 253379A and the Approved Flight Manual contains General Amendment No. G1. The Approved Flight Manuals are document HS.1.19 for FAA certified aircraft and document HS.1.18 for UK CAA certified aircraft.

NOTE 39. Modification 253410A introduces aerodynamic improvements to the tailplane/elevator configuration. BAe.125 Series 1000 and Hawker 1000 aircraft modified as specified above are to be operated in

accordance with the Approved Flight Manual Document No. HS.1.19 containing General Amendment No. G6.

NOTE 40. Modification 259976 Part A introduces the BAe.125 Series 800A (U-125) aircraft intended for Airborne Flight Inspection Operations. These aircraft embody features which would not normally be found on Civil Transport Aircraft including various provisions to enable the aircraft to be modified under STC action in the USA. Where these provisions have been made for installations of equipment under STC action, these have been shown to comply with the associated installation requirements and be of no hazard to the aircraft, but have not been investigated for their intended function with installation of any STC.

A BAe.125 Series 800A aircraft modified as specified above must be operated in accordance with the Approved Flight Manual Document No. HS 1.16 containing Particular Amendment No. P 60.

NOTE 41. Modification 253686A introduces the Hawker 1000 designation and makes the requisite changes to identification plates and the limitations placard. This change is reflected in the Approved Flight Manual Document No. HS.1.19 containing Particular Amendment No. P 34. The Hawker 1000 is only a name change from the former BAe.125 Series 1000A. All Service Information published for the BAe.125 Series 1000A is equally applicable to the Hawker 1000.

NOTE 42 Modification 253558A introduces the Hawker 800 designation and makes the requisite changes to identification plates and the limitations placard. This change is reflected in the Approved Flight Manual Document. No. HS 1.16 containing Particular Amendment P 63. The Hawker 800 is only a name change from the former BAe.125 Series 800A. All Service Information published for the BAe.125 Series 800A is equally applicable to the Hawker 800.

NOTE 43. Modification 253650A introduces the PW305B engine. Embodiment of this modification changes the V_{MC} (Minimum Control Speed) limits. A BAe.125 Series 1000A, 1000B or Hawker 1000 aircraft modified as specified above is to be operated in accordance with the Approved Flight Manual Document No. HS.1.19 containing Particular Amendment No. P 17.

NOTE 44. The maximum zero fuel weight on the BAe.125 Series 1000A, 1000B and the Hawker 1000 may be increased to 20,400 lbs. provided that a revised label is installed in accordance with Modification 25A714A and the Approved Flight Manual HS.1.19 containing Particular Amendment No. P 33.

NOTE 45. An optional Modification Number 253608A deletes the external baggage door on the BAe.125 Series 1000A, 1000B and Hawker 1000 aircraft.

NOTE 46. The maximum Zero Fuel Weight for the Models DH.125 Series 3A/RA and the HS.125 Series 3B/RA may be increased to 14,700 lbs. provided that Modification 25A767A is embodied and the Approved Flight Manual HS.1.3. contains Particular Amendment No. P 89. V_{MO} is also reduced.

NOTE 47. Modification 253564A with associated changes introduces the model Hawker 800XP. The Garrett AiResearch TFE 731-5R Turbofan engines originally fitted to the Hawker 800 airplanes are replaced by Allied Signal Engines TFE 731-5BR. In addition to the installation of the TFE 731-5BR engines, the above modification also introduces the following changes:

i) Dee Howard TR5000BR thrust reversers fitted as standard.
ii) Increase in certificated ramp, take-off and maximum zero fuel weights.
iii) Vortilons replace wing fences and Hawker 1000 aileron servo tab gearing is introduced.
iv) Rudder Bias moment arm is reduced to 2.72".
v) Mach Trim System is fitted.
vi) 3 Wheel ECS is fitted as standard.
vii) 38 liter TKS tank is fitted.
viii) A Hawker 800XP airplane is to be operated in accordance with the Approved Flight Manual, Document No. HS.1.22. with appropriate Particular Amendments.

NOTE 48 UK CAA has made an assessment that all mandatory actions are contained in the instructions for Continued Airworthiness as well as embodied during the production of the Hawker model airplanes. This is documented in UK CAA letter reference 9/33/3956/A 24890 dated July 26, 1995.

NOTE 49. Modification 25B047A introduces the Hawker 800 intended for operation by the Japan Air Self Defense Force as a U-125A aircraft. Document CJE.CPD.D.272.001381 Issue 1, outlines the changes made to a standard Hawker 800 aircraft to achieve the delivery standard exported from the manufacturer. This modification was approved by the UK CAA on December 7, 1994 and is accepted by FAA as having demonstrated compliance with the particular requirement of the customer. Where provisions have been made for the fitment of equipment by the Japanese customer, these have been shown to comply with the associated installation requirements and be of no hazard to the aircraft, but have not been investigated for their intended function.

A Hawker 800 aircraft modified as specified above must be operated in accordance with the Approved Flight Manual Document. No. HS.1.16 containing Particular Amendment No. P 64 and any other applicable approved amendments.

NOTE 50. Some aircraft delivered new from the UK to international customers may not necessarily comply in full with the defined certification basis on which this TC has been granted due to overriding Foreign Authority requirements which have been satisfied for aircraft delivered into their country.

There are two basic certification standards for the DH/HS/BH/BAe.125/Hawker series of airplanes. One is the US FAA Type Certificate standard. Aircraft certified to this standard are identified with an “A” in the Model suffix. The other certification standard is based on requirements established by the UK Civil Aviation Authority (CAA). Aircraft certified to the UK CAA standards are identified as “B” versions and include the following Models, Series: 1B, 1B-522, 1B/S-522, 1B/R-522**,** 3B, 3B/R, 3B/RA, 3B/RB, 3B/RC, F3B, F3B/RA, 400B, 400B/1, 401B, 403A(C), 403B, F400B, F403B, 600B, 600B/1, 600B/2, 600B/3, F600B, 700B, 800B, and 1000B. The ‘B’ models are equivalent to the ‘A’ models and meet U.S. certification requirements with the exception of the overriding UK CAA requirements and customer requested optional modifications approved by the UK CAA. While most countries outside of the UK and USA accept either “A” , “B”, or both versions of the aircraft, modifications of these aircraft are sometimes required to satisfy national variations in the certification standards established by the importing countries. Due to the wide range of potential configurations, specific instructions for modifying an airplane from one country standard to another are not available in a pre-published format. In those cases where it does become necessary to convert an aircraft from one certification standard to another, or to show the equivalency to the U. S. standard, the document used will be a serial number specific Service Bulletin issued by the Type Certificate Holder. This Service Bulletin will be FAA Approved.

NOTE 51. Raytheon Aircraft Company Service Bulletin No. 00-11 titled "General-Record of UK Airworthiness Directives (AD) at the Time of Transfer of ICAO Annex 8 Responsibilities from UK CAA to US FAA" will be used to document the AD’s issued by the UK CAA prior to August 1, 1995.

NOTE 52. Regulatory requirements applicable to HS.125 Series 1B, 1B-522, 1B/R-522, 1B/S-522, 3B, 3B/R, 3B/RA, 3B/RB, 3B/RC, F3B and F3B/RA United Kingdom (UK) certified aircraft ("B" aircraft) to be eligible for U. S. FAA Transport category airworthiness certificates and registration as equivalent to an "A" aircraft are: (i) UK Air Registration Board V. A. Note 1, Issue 1, dated 19 April, 1961, (ii) Modification 251265 or equivalent for stall warning when the throttles are opened while lift dump or air brakes are extended, (iii) Modification 255051 or equivalent for passenger oxygen systems, (iv) Modification 251266 or equivalent for a speed warning device set in accordance with the requirements of FAA S.R. 450A, (v) FAA Exemption Number 573 grants exemption from CAR 4 b.437 Fuel Jettisoning System, and (vi) See Requirements of NOTE 50.

NOTE 53. Regulatory requirements applicable to HS.125 Series 400B, 400B/1, 401B, 403B, 403A(C), F400B and F403B United Kingdom (UK) certified aircraft ("B" aircraft) to be eligible for U. S. FAA Transport category airworthiness certificates and registration as equivalent to an "A" aircraft are: (i) UK Air Registration Board V. A. Note 1, Issue 1, dated 19 April, 1961, (ii) Modification 251265 or equivalent for stall warning, (iii) Modification 255051 or equivalent for passenger oxygen systems, (iv) Modification 251266 or equivalent for a speed warning horn set in accordance with the requirements of FAA S.R. 450A, and (v) See requirements of NOTE 50.

NOTE 54. Regulatory requirements applicable to HS.125 Series 600B, 600B/1, 600B/2, 600B/3 and F600B United Kingdom (UK) certified aircraft ("B" aircraft) to be eligible for U. S. FAA Transport category airworthiness certificates and registration as equivalent to an "A" aircraft are: (i) UK Air Registration Board V. A. Note 1, Issue 1, dated 19 April, 1961, (ii) Modification 251721 or equivalent for stall warning, (iii) Modification 256263A or equivalent for passenger oxygen systems, (iv) Modification 252261A or equivalent for a speed warning horn to meet compliance with FAR 25.1303 (c) (1) , (v) If applicable meet NOTE 18, and (vi) See requirements of NOTE 50.

NOTE 55. The models HS.125 Series 600B/1, 600B/2 and 600B/3 were aircraft that had been exported to various countries and modified to operate within that countries Certification Agencies rules. These aircraft were later exported to the United Kingdom and inspected and modified to operate equivalent to a HS.125 Series 600B aircraft. To be eligible for U.S. FAA Transport category airworthiness certificate and registration these aircraft will be considered a model HS.125 Series 600B and shall meet the requirements of NOTE 54.

NOTE 56. Regulatory requirements applicable to HS.125 Series 700B United Kingdom (UK) certified aircraft ("B" aircraft) to be eligible for U. S. FAA Transport category airworthiness certificates and registration as equivalent to an "A" aircraft are: (i) UK Air Registration Board V. A. Note 1, Issue 1, dated 19 April, 1961, (ii) Modification 252509 or equivalent for stall warning, (iii) Modification 252036A or equivalent for passenger oxygen systems, (iv) Modification 252523 or equivalent for a speed warning horn to meet compliance with FAR 25.1303 (c) (1), (v) Aircraft must be operated using Approved Flight Manual Document HS.1.9 with the appropriate Particular Amendments, and (vi) See requirements of NOTE 50.

NOTE 57. Regulatory requirements applicable to BAe.125 Series 800B and some Hawker 800 that were United Kingdom (UK) certified aircraft ("B" aircraft) to be eligible for U. S. FAA Transport category airworthiness certificates and registration as equivalent to an "A" aircraft are: (i) UK Air Registration Board V. A. Note 1, Issue 1, dated 19 April, 1961, (ii) Modification 252509 or equivalent for stall warning, (iii) Modification 252036A or equivalent for passenger oxygen systems, (iv) Modification 252523 or equivalent for a speed warning horn to meet compliance with FAR 25.1303 (c) (1), (v) Aircraft must be operated using Approved Flight Manual Document HS.1.16 with the appropriate Particular Amendments, and (vii) See requirements of NOTE 50.

NOTE 58. Regulatory requirements applicable to BAe.125 Series 1000B and some Hawker 1000 that were United Kingdom certified aircraft ("B" aircraft) to be eligible for U. S. FAA Transport category airworthiness certificates and registration as equivalent to an "A" aircraft are: (i) UK Air Registration Board V. A. Note 1, Issue 1, dated 19 April, 1961, (ii) Modification 252509 or equivalent for stall warning, (iii) Modification 252036A or equivalent for passenger oxygen systems, (iv) Modification 252523 or equivalent for a speed warning horn to meet compliance with FAR 25.1303 (c) (1), (v) Aircraft must be operated using Approved Flight Manual Document HS.1.19 with the appropriate Particular Amendments, and (vii) See requirements of NOTE 50.

NOTE 59. The Hawker 800XP aircraft was designed to meet this Type Certificate standard and receive the U. S. FAA Transport Category Airworthiness Certificates. Some individual aircraft may not necessarily comply in full with the defined certification basis on which this TC has been granted due to overriding Foreign Authority requirements which have been satisfied to deliver in their country. Any airplane returning to the U.S. and requesting a U.S. FAA airworthiness certificate and registration must be modified to remove the exporting countries modifications to meet this Type Certificate standard. The aircraft must be operated using Approved Flight Manual Document HS.1.22 with the appropriate Particular Amendments. Contact the Type Certificate Holder for information regarding required changes to specific serial numbered airplanes returning to the U.S. See requirements of NOTE 50.

NOTE 60. The following serial numbered aircraft were manufactured in the UK by Raytheon Corporate Jets, Inc.

Hawker 800 (U-125A) 258245, 258247 and 258250
Hawker 800 258255 through 258265, 258267, 258269 through 258276
Hawker 1000 259043 through 259047.

Aircraft manufactured in the UK by Raytheon Corporate Jets, Inc. under license to Raytheon Aircraft Company can be identified in NOTE 6 (d).

NOTE 61. Some aircraft were manufactured and delivered to the United States using only a North American (NA) reference number on the aircraft data plate. Service Bulletin SB.00-12 provides a cross reference listing of the North American (NA) reference numbers against serial numbers (25XXX or 25XXXX).

NOTE 62. The contents of the ventral fuel tank are reduced by 4.8 gallons for aircraft which have fitted an external toilet servicing facility.

NOTE 63. The following serial numbered Hawker 800 (U-125A), Hawker 800XP and Hawker 1000 aircraft were manufactured by Raytheon Aircraft Company in the U. S.:

Hawker 800 (U-125A) and Hawker 800XP: 258297, 258301, 258304, 258306, 258309, 258311, 258313, 258315, 258317, 258319, 258320, 258322, 258325, 258326, 258331, 258333, 258334, 258336 and 258338 and on.
Hawker 1000: 259003

NOTE 64. The BAe.125 Series 800A, Hawker 800 and Hawker 800XP have been approved for Reduced Vertical Separation Minimum (RVSM) flight. If the aircraft has a Honeywell Avionics Package, modification 25F731A must be embodied. If the aircraft has a Collins Avionics Package, modification 25F731B must be embodied.

The BAe.125 Series 1000A and the Hawker 1000 have been approved for Reduced Vertical Separation Minimum (RVSM) flight. Modification 25F856A must be embodied with the Honeywell Avionics Package.

Final certification for RVSM operations must be obtained by the operator from the local FAA Flight Standards District Office (FSDO).

NOTE 65. Master Drawing List 800E0165 introduces the FAA approved modifications for the Hawker 800XP aircraft for operation by the Brazilian Air Force for Airborne Flight Inspection Operations. A Hawker 800XP aircraft modified as specified above must be operated in accordance with the Approved Flight Manual Document No. HS 1.22 containing Supplement 9, Issue 2. The following serial numbered aircraft were modified per the above master drawing list 258401, 258421, 258434 and 258447.

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